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AMBIENT SCATTERING FROM RING-SYMMETRIC
SPACECRAFT EXHAUST PLUME

by

Joseph Falcovitz

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ABSTRACT

We present a first-collision model for the evaluation of return flux from the exhaust plume of a ring-symmetric HF/DF laser in LEO, generated by an incident flux of ambient molecules traveling at orbital speed. The steady plume is bounded by a pair of lip-centered rarefaction fans, and unless spacecraft attitude enables incident air molecules to reach the plume through the cavitation regions that extend beyond these fans, the spacecraft is shielded from ambient scattering by its own plume. Assuming hard-spheres collisions, the first-collision model is given by a simple closed-form expression that can be regarded as a source term for scattered exhaust molecules. This source term is integrated numerically throughout the fan, yielding the flux arriving at some surface "target point". Quantitatively, it is shown that for a typical HF/DF laser exhaust the contamination level generated by ambient scattering is not significant. It was found that the maximum return flux of HF+DF constitutes about 2% of the incident ambient flux; this ratio will be nearly constant for LEO altitudes. The value of this flux ratio is shown to be dependent on the molecular collision model; it may change upon replacing the hard-spheres approximation by a more realistic collision model. A possible modification of spacecraft charging by the exhaust was examined, including production of HF^- and DF^- . The only significant effect seemed to be shadowing of the downstream half of the spacecraft at oblique orbital attitudes.

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TABLE OF CONTENTS

1.	INTRODUCTION.....	1
2.	COMPUTATION OF THE PLUME FLOW FIELD.....	4
3.	AMBIENT SCATTERING.....	8
3.1	First Collision Model.....	8
3.2	Flux Integration Scheme.....	10
4.	RESULTS AND DISCUSSION.....	17
5.	SPACECRAFT CHARGING.....	23
6.	CONCLUDING REMARKS.....	27
7.	REFERENCES.....	29
	APPENDIX A. DESCRIPTION OF AMB CODE.....	31
A.1	Description of Subroutines.....	31
A.2	Listing of AMB code.....	34
8.	DISTRIBUTION LIST.....	52

LIST OF FIGURES

Figure 1-1.	Ring-Symmetric HF/DF Laser Exhaust Plume.....	3
Figure 1-2.	Schematic Description of Ambient Scattering. The Cavitation Region is Bounded by Lines CA and CB.....	3
Figure 2-1.	Power $\delta(0,\beta)$ for the power-law Approximation.....	6
Figure 2-2.	Variation of Mach Number along Characteristic Line $\beta = 13$	7
Figure 3-1.	Incidence-Plane Description of Flux Integration Scheme.....	14
Figure 3-2.	Hard-Spheres Collision Notation.....	15
Figure 3-3.	Scattering Envelope for Hard-Spheres Collision.....	16
Figure 4-1.	Variation of Return Flux with Target Point (X_s). Target Point at Incidence-Plane ($\varphi_A = 0$) and Constant Incidence-Angle ($\psi_A = 20^\circ$).....	20
Figure 4-2.	Variation of Return Flux with Ambient Incidence Angle (ψ_A). Fixed Target Point ($X_s = 1$ m) Located at Incidence-Plane ($\varphi_A = 0$).....	21

Figure 4-3. Variation of Return Flux with Ambient Azimuth Angle (φ_A).
Fixed Target Point ($X_s = 1$ m) and Ambient Incidence Angle ($\psi_A = 20^\circ$)..... 22

LIST OF TABLES

Table 4-1. Typical Operating Conditions of HF/DF Laser Exhaust..... 19

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1. INTRODUCTION

This presentation is part of a study on the gas dynamics of ring-symmetric exhaust plumes in space, conducted at the Naval Postgraduate School in Monterey. A ring-symmetric jet has zero thrust, which makes it suitable as an exhaust configuration for various open loop power plants designed to produce high power for relatively short durations. One such system is an envisioned space-based chemical laser, shown schematically in Fig. 1-1. In the case of a chemical laser, a ring-symmetric configuration would also enable the laser radiation to emerge in the form of an axisymmetric beam.

The exhaust nozzle should be designed to bring the outgoing flow to a supersonic speed at the nozzle exit surface. The near field of a free jet is then composed of an inner core bounded by a pair of ring-symmetric rarefaction fans centered at the nozzle lips (Fig. 1-1). Beyond the limiting characteristic surface of the centered rarefaction waves (CRW), a near-vacuum condition prevails. For the purpose of continuum gas dynamic analysis, we assume it is a perfect vacuum.

An earth orbiting vehicle is subject to an oncoming stream of ambient molecules at a speed of $U_A \approx 8$ (km/sec), in a direction depending upon its orientation relative to the orbital velocity vector. This speed is sufficiently high to cause backscattering of exhaust molecules (see schematic description in Fig. 1-2) moving at speeds appropriate to chemical combustion (about 2 to 4 km/s). However, large exhaust plumes, having achieved stationary flow, may be sufficiently dense at their outer fringes to effectively trap and entrain all oncoming ambient molecules. Thus, ambient scattering may be significant only in selected ranges of attitude angles, at which ambient molecules can reach the vicinity of the spacecraft by traveling almost collisionlessly through cavitation regions. Exhaust molecules that may be "candidates" for ambient scattering will hence come from plume segments flanked by cavitation regions. The contribution of ambient scattering to contamination will thus be highly dependent upon spacecraft geometry and orientation. This may well affect spacecraft design and operating procedures.

The purpose of this report is to present a first-collision model for estimating the flux of exhaust molecules backscattered from the fringes of the plume by ambient molecules, along with results of sample flux computations performed on a typical HF/DF laser exhaust configuration. The flow field throughout the plume is assumed to be governed by the equations of continuum gas dynamics. In principle, the flow could be obtained by solving the governing equations, i.e., the equations for stationary isentropic flow in two-dimensional axisymmetric coordinates. In practice, this is normally

accomplished by integrating the flow equations in characteristic form, using some finite difference scheme (method-of-characteristics). We have performed such computations, but given the complexity of applying them to the subsequent integration of ambient scattering flux (due to the need for two-dimensional interpolations from an irregular solution grid), we opted for a different alternative : a closed-form approximation to the ring-symmetric CRW, based on an analytic expression for flow variables along characteristic lines that fan out from the nozzle lip.

The plan of this report is as follows. In Ch. 2 we outline the approximation to the ring-symmetric CRW and present some computation results that demonstrate its accuracy. In Ch. 3 we describe the first-collision model and the 3-D spatial integration scheme for computing the flux arriving at the cylindrical spacecraft. In Ch. 4 some results of backscattered flux of corrosive molecules (HF + DF), showing flux variation with target point location (X_s) and attitude angles (ψ_A, ϕ_A) are presented. In Ch. 5 we take up the subject of spacecraft charging, using results of ambient scattering to assess the effect of laser exhaust on spacecraft charging. This is followed by concluding remarks in Ch. 6 and a list of references in Ch. 7. A concise description of the flux computation code "AMB" is given in Appendix A, followed by the code listing.

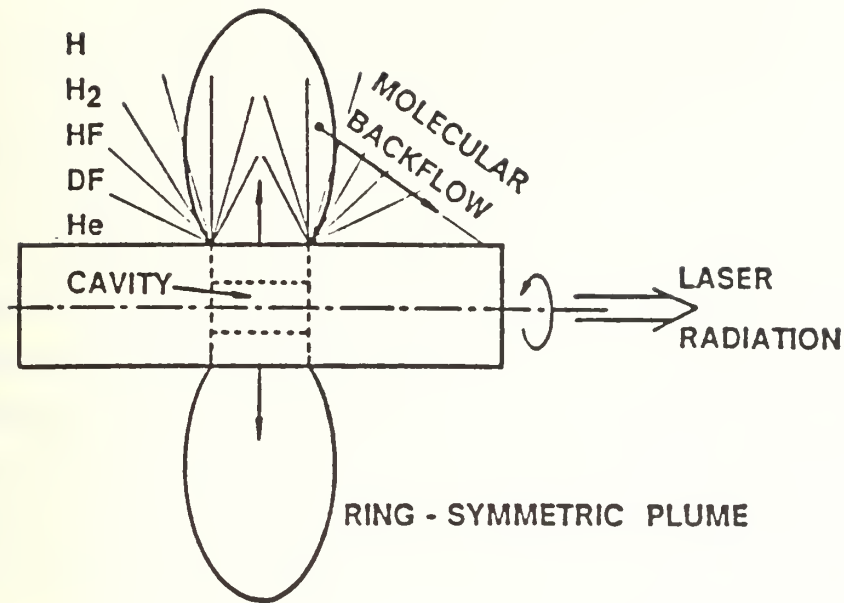


Figure 1-1. Ring-Symmetric HF/DF Laser Exhaust Plume.

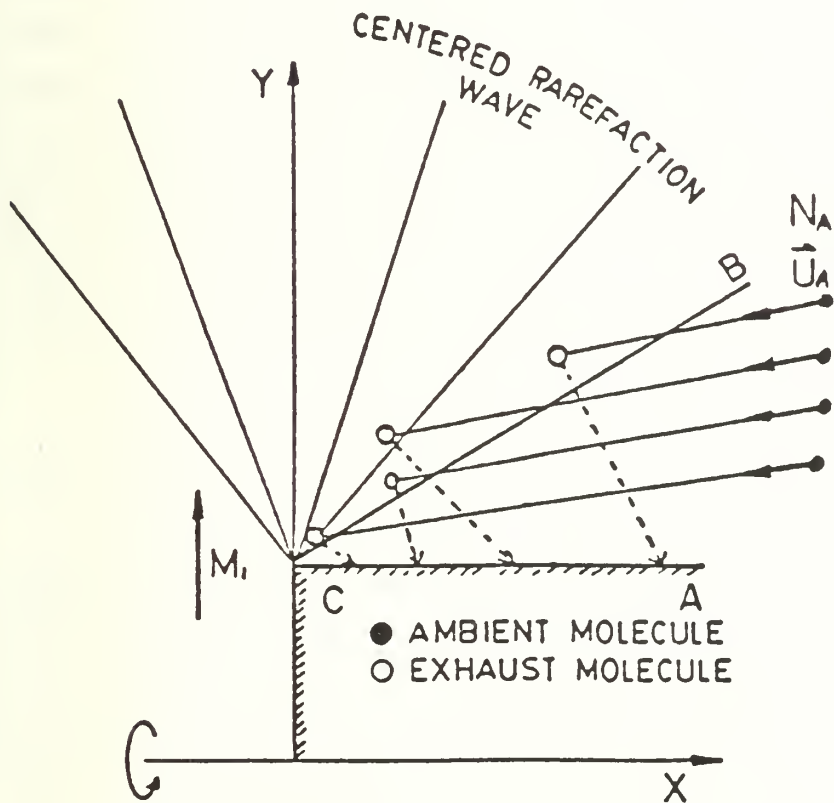


Figure 1-2. Schematic Description of Ambient Scattering. The Cavitation Region is Bounded by Lines CA and CB.

2. COMPUTATION OF THE PLUME FLOW FIELD

Most ambient molecules entering the CRW that flanks the exhaust plume are stopped within several mean free paths from their point of entry. A quantitative estimate of ambient back-scattering would thus depend on the flow field at the outer (hypersonic) fringes of the lip-centered CRW. Even though the flow in those regions is generally past the surface of continuum breakdown, the density there is reasonably well approximated by the continuum flow field, as demonstrated by Bird's Monte-Carlo simulation of a Prandtl-Meyer expansion to vacuum [1]. The evaluation of ambient scattering thus calls for an ancillary computational procedure capable of rendering the continuum flow field at a large number of points in the ring-symmetric CRW of an exhaust plume. This method was described in a recent report [2]. Here we just outline the key ideas and main results of this approximation method.

Our analytic approximation to a ring-symmetric CRW is formulated as follows. In a planar CRW (Prandtl-Meyer flow) all flow variables are uniform along the characteristic lines that fan out from the corner (we assume they are the C^+ family). In the ring-symmetric case the flow near the corner approaches asymptotically a corresponding planar CRW flow, which we term the *associate* CRW. However, the gradients along C^+ characteristics at the corner of a ring-symmetric CRW do not vanish as in a planar CRW. The key idea is thus: evaluate flow gradients in C^+ directions at the corner, then use them to extrapolate the associate CRW along C^+ lines to a finite distance from the corner. The extrapolation is a nonlinear function of the radial coordinate y , chosen so that the ensuing expression conforms exactly to the flow at the leading (exit) characteristic $C^+(\beta_1)$. Omitting all details of the analysis, the resulting approximation is presented as the following power-law:

$$f(\alpha, \beta) = f(0, \beta) \left[y(\alpha, \beta) / y(0, \beta) \right]^{\delta(0, \beta)} \quad (2-1)$$

where f is the streamtube area ratio for isentropic flows ($f=1$ at a sonic point), β is the Mach number of a particular characteristic line at the corner, α is a coordinate along the $C^+(\beta)$ characteristic line ($\alpha=0$ at the corner), and y is the radial coordinate of a point on the characteristic line $C^+(\beta)$. The Mach number at point (α, β) is readily determined from $f(\alpha, \beta)$ using the standard relation between area ratio and Mach number [3]. A closed-form expression for $\delta(0, \beta)$ was developed but is not given here; instead, this function is shown in Fig. 2-1. We note that δ approaches the asymptotic value of $2/(3-\gamma)$ as β increases to infinity, and that generally $1 < \delta(0, \beta) < 2$ so that streamtubes diverge at a rate intermediate between that of cylindrical and spherical expansion flows.

Clearly, in an isentropic flow all thermodynamic variables, and in particular density, can be evaluated from f . This approximation is readily applied to the hypersonic portions of a ring-symmetric CRW since it turns out that characteristic lines are nearly straight there, which means that the characteristic line $C^+(\beta)$ passing through a given point can be readily determined. As a demonstration of the degree of accuracy obtainable from this approximation, we show in Fig. 2-2 the variation of Mach number along a characteristic line in the ring-symmetric CRW, compared with an accurate method-of-characteristics computation. This comparison demonstrates that the analytic approximation is reasonably accurate to nearly ten corner-radii away from the corner.

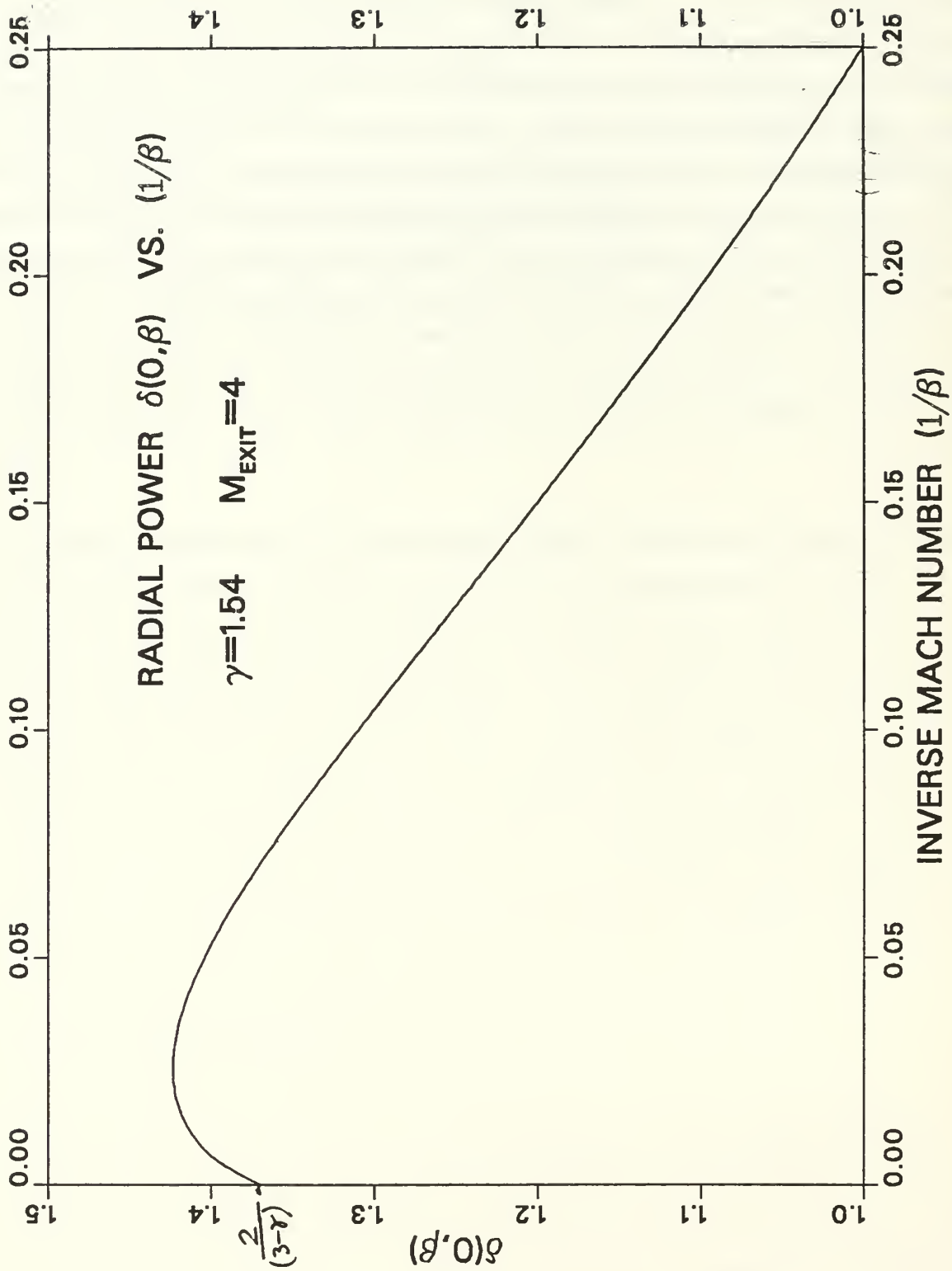


Figure 2-1. Power $\delta(0,\beta)$ for the power-law Approximation.

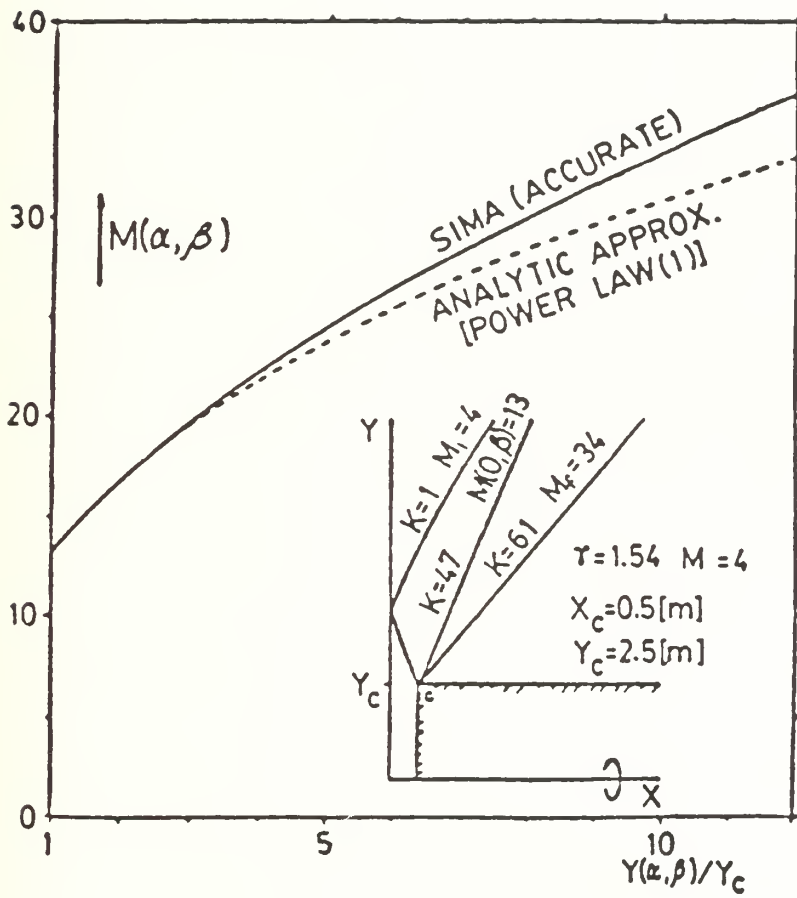


Figure 2-2. Variation of Mach Number along Characteristic Line $\beta = 13$.

3. AMBIENT SCATTERING

When a rocket or laser exhaust is released into space from an earth-orbiting spacecraft, it encounters an oncoming stream of ambient molecules flowing at the orbital speed of $U_A \approx 8$ (km/sec). At altitudes higher than 200 (km), the air/air mean free path exceeds 250 (m), so that it is considerably larger than almost any spacecraft. Consequently, ambient molecules would hardly be subjected to a shock transition prior to their impact at the spacecraft or exhaust plume. In this chapter we describe the formulation of the first-collision model in Section 3.1 and then proceed to present the derivation of the flux integration scheme for hard-sphere collisions in Section 3.2.

3.1 First Collision Model

The highest ambient number density that we consider for earth-orbiting spacecrafts is $N_A = 1 \times 10^{16}$ (m^{-3}), which roughly corresponds to Sunspot Maximum at 200 (km) [4]. The typical laser exhaust (Table 4-1) would reach a number density of about 2×10^{19} (m^{-3}) at the very high Mach number of 30. Hence, ambient flux constitutes just a slight perturbation to the near-field portion of a typical laser exhaust plume. Obviously, ambient molecules that penetrate the plume, would subsequently be entrained by the main flow. But how far do they penetrate? And would exhaust molecules scattered by them reach the spacecraft? In seeking answers to these questions, we are led to some interesting observations concerning ambient scattering.

Consider the HF laser depicted in Fig. 1-1. The spacecraft diameter is 5 (m) and the centrally located ring-symmetric nozzle is 1 (m) wide. Typical operating conditions (Table 4-1) are assumed. They are based on some experimental HF/DF laser studies conducted at TRW [5,6]. Suppose that the spacecraft axis is normal to the orbital velocity vector (normal incidence). Let the plane of incidence be the plane defined by the intersection of the spacecraft axis with the orbital velocity vector. The probability that an ambient molecule traveling in the plane of incidence would reach the spacecraft collisionlessly is $\exp(-\eta)$, where η is its expected number of collisions with exhaust molecules. We define the number η as "molecular thickness", in analogy to "optical thickness". So in order to determine the extent to which ambient molecules at normal incidence reach the spacecraft, we seek the distribution of radial molecular thickness as function of distance from the spacecraft mid-plane (normal to axis at its midpoint).

For this purpose we computed the ring-symmetric exhaust flow field, using a semi-inverse marching characteristics scheme [7]. The marching was in the radial direction, starting with uniform flow at the nozzle exit; the computation was carried on until it became evident that even at a distance of 20 (m) from the mid-plane, the radial molecular thickness was well over 40. The entire spacecraft was thus shielded from any ambient scattering at (or near) normal incidence. This shielding effect has two significant implications which we discuss briefly below.

- (a) It is present only during stationary exhaust flow. At startup and shutdown phases, ambient scattering may be substantial even at normal incidence.
- (b) During the stationary phase, ambient scattering is substantial only at attitude angles that enable ambient molecules to reach the vicinity of the plume by traveling through "molecularly thin" cavitation regions that flank the plume. We thus anticipate a decisive dependence of ambient scattering on attitude variations, whenever those variations steer the spacecraft into or out of a shielded posture.

As a first attempt at a quantitative estimate of ambient scattering flux, we have formulated a simple first-collision model of this effect. In the sequel we present an outline of the model, along with some sample results evaluated for an HF laser configuration identical to that considered for the shielding effect mentioned above.

The basic idea is the following. Ambient molecules entering an exhaust plume, require several collisions to become fully "accommodated" with the main flow (i.e., to be entrained by the main flow at the prevailing flow velocity and temperature). One may reasonably approximate this process by considering just one collision - the first.

With the help of some additional assumptions, we were able to derive a closed form expression for the flux of exhaust molecules that arrive at the spacecraft following a first collision with an ambient molecule. The main assumptions of this model are :

- (1) **FIRST COLLISIONS:** Only first collisions for either ambient or exhaust molecules are considered. Hard-spheres elastic collisions are assumed. Upon a second collision of either an ambient or an exhaust molecule, it is considered "lost" (i.e., it joins the main flow). Collisions of ambient molecules with spacecraft surfaces are ignored. Ambient molecules are assumed to traverse cavitation regions collisionlessly.

- (2) COLD FLOW: The oncoming ambient air flow is deemed "cold"; i.e., all molecules move at the uniform orbital velocity. The same "cold" assumption is applied to the exhaust flow, since most ambient scattering takes place at plume regions of very high Mach numbers (well over 10, in the present case).
- (3) CRW Flow Field: ring-symmetric CRW flow field is determined from the power-law approximation described in Ch. 2 above. This approximation approaches Prandtl-Meyer flow at points whose distance from the nozzle lip is much smaller than the spacecraft radius.

Based on these assumptions, ambient scattering is represented as a source term for side-scattered exhaust molecules, distributed throughout the lip-centered rarefaction fan. The total flux arriving at a specified point on the cylindrical spacecraft is readily computed by integrating numerically that source distribution over the entire ring-fan.

The highlights of the spatial integration scheme (Fig. 3-1) are as follows. The limiting characteristic surface ($M = \infty$) of the ring-symmetric CRW is divided into surface elements formed by dividing the surface into a set of ring-strips which are subdivided in the circumferential (azimuthal) direction (ϕ) into surface elements. The line-of-sight ($\vec{\Omega}$) from the "target point" on the spacecraft to the center of each surface element is extended into the ring-symmetric CRW, and flux integration using the first-collision source term with appropriate weight factors is performed along this line until convergence is attained. Contributions from each surface element are summed, taking care to disregard portions of the ring-symmetric CRW that are shadowed by the cylindrical spacecraft (either the line-of-sight or the trajectory of oncoming ambient molecules may be shadowed). Some further details of the flux integration scheme and hard-spheres collisions are provided in Section 3.2 below.

3.2 Flux Integration Scheme

The description of the first collision model is hereby supplemented with an outline of the expressions used in the flux integration and their derivation. The integration scheme for flux arriving at point X_s on the spacecraft is depicted in Fig. 3-1. Note that only the plane of incidence is shown in Fig. 3-1; at other azimuth angles the geometry is not co-planar, so 3-D geometrical expressions are used to get the coordinates (ψ, ϕ and radial distance $(y^2 + z^2)^{1/2}$) from $\vec{\Omega}$ and \mathbf{S} ; the derivation of these geometrical relations is straightforward, so that we omit these details in the present report. The total number flux $Q_i(X_s)$ of i exhaust molecules arriving at point X_s is given by the following expression :

$$Q_i(X_s) = \int d^3\vec{\Omega} \cos\alpha_s \sum_k \int_0^\infty dS \sigma_{ik} h_i N(S) h_k N_A |\vec{U}(S) - \vec{U}_A| \exp[-\eta_k(S)] P_{ik}(S, -\vec{\Omega}) \exp[-\eta_{ik}(S)]$$

$$\eta_k(S) = \sum_i \int_0^{t(S)} dt' \sigma_{ik} h_i N(t') |\vec{U}(t') - \vec{U}_A| / |\vec{U}_A| \quad (3-1)$$

$$\eta_{ik}(S) = \sum_j \int_0^S dS' \sigma_{ij} h_j N(S') |\vec{U}_{ik}(S) - \vec{U}(S')| / |\vec{U}_{ik}(S)|$$

$()_i$ $()_j$ - Exhaust species

$()_k$ - Ambient species

These expressions are interpreted as follows. The collision depicted in Fig. 3-1 is between exhaust molecule m_i and ambient molecule m_k . The exhaust molar fractions h_i and ambient molar fractions h_k are assumed uniformly constant, and so are the ambient velocity \vec{U}_A and number density N_A . The exhaust velocity $\vec{U}(S)$ and number density $N(S)$ are function of the location in the flow field defined by $\vec{\Omega}$ and S . These flow variables are computed by first evaluating the coordinates of point $\vec{\Omega}, S$ (Fig. 3-1) in the ring-symmetric CRW from the 3-D geometry, and then employing the power-law approximation outlined in Ch. 2 above, to get all flow variables for a ring-symmetric CRW. In this computation we exploit the fact that characteristic lines fanning out from the nozzle lip are nearly straight lines at the low pressure side of the ring-symmetric CRW.

The $\vec{\Omega}$ integration is performed numerically according to the scheme outlined in Section 3.1 above, as a summation over elements of solid angle ($\Delta^3\vec{\Omega}$) subtended by area elements on the limiting characteristic cone ($\psi = \psi_f$).

The S integration is considerably more complex. The integrand for this integration is derived as follows. Denote by L the line-of-sight distance between point X_s and fan point $\vec{\Omega}, S$. A volume element at the fan point is given by $\Delta v = L^2 \Delta S \Delta^3\vec{\Omega}$. The number of ik pair collisions in Δv per unit time is $\sigma_{ik} h_i N(S) h_k N_A |\vec{U}(S) - \vec{U}_A| \exp[-\eta_k(S)] \Delta v$, where $\eta_k(S)$ denotes the expected number of collisions of ambient molecule k with any exhaust molecule, between its point of entry into the plume and point $\vec{\Omega}, S$. We now multiply this term by $\exp[-\eta_{ik}(S)]$ which is the probability that exhaust molecule i scattered by ambient molecule k would travel from point $\vec{\Omega}, S$ to point X_s collisionlessly, where $\eta_{ik}(S)$ is the expected number of collisions for this path segment. (Note that in Eq. (3-1) the summation in the expression for $\eta_{ik}(S)$ is over all exhaust species j).

The final step in constructing the integrand for the \mathbf{S} integration involves the post-collision directional distribution function $P_{ik}(\mathbf{S}, -\vec{\Omega})$, whose derivation will be given in the sequel. We multiply the integrand by $P_{ik}(\mathbf{S}, -\vec{\Omega}) \Delta^3 \vec{\Omega}_c$ which is the fraction of i exhaust molecules scattered by k ambient molecules into a solid angle element $\Delta^3 \vec{\Omega}_c$ about the unit vector $-\vec{\Omega}$. Considering the flux arriving at a surface area element ΔA_s around point \mathbf{X}_s , the solid angle element subtended by ΔA_s is $\Delta^3 \vec{\Omega}_c = \Delta A_s \cos \alpha_s / L^2$. Eq. (3-1) for $Q_i(\mathbf{X}_s)$ now follows upon dividing the resulting expression by ΔA_s , thus referring the arriving flux to a unit area at the point of arrival \mathbf{X}_s .

Numerically, the \mathbf{S} integration was performed using the classical Runge-Kutta scheme (fourth order). The integration for $\eta_{ik}(\mathbf{S})$ and $\eta_k(\mathbf{S})$ has to be repeated at each point \mathbf{S} . We found reasonable convergence with 4 points in the $\eta_k(\mathbf{S})$ integration and 6 points in the azimuth integration. The \mathbf{S} integration was terminated when convergence was attained (this is the meaning of the upper limit ∞ in the \mathbf{S} integral in Eq. (3-1)). The summation over new strips on the limiting cone ($\psi = \psi_f$) was also terminated upon convergence. The CPU time consumed per target point was about 100 (sec) on IBM 3033 mainframe.

We now take up the derivation of an expression for the post-collision directional distribution function $P_{ik}(\mathbf{S}, -\vec{\Omega})$, which we denote hereafter as $P(-\vec{\Omega})$. We adopt the pair-collision notation presented in Fig. 3-2 for the hard-sphere collision analysis.

As a consequence of conservation of momentum and energy (elastic collisions), the center-of-mass velocity \vec{C}_m and the magnitude of the relative velocity \vec{C}_r are unchanged by the collision [8]. The post-collision velocities are given by :

$$\begin{aligned}
 \vec{C}_1^* &= \vec{C}_m + \mu_2 \vec{C}_r^* & \vec{C}_2^* &= \vec{C}_m - \mu_1 \vec{C}_r^* \\
 \vec{C}_r &= \vec{C}_1 - \vec{C}_2 & \vec{C}_r^* &= \vec{C}_1^* - \vec{C}_2^* \\
 \mu_1 &= m_1 / (m_1 + m_2) & \mu_2 &= m_2 / (m_1 + m_2) \\
 \vec{C}_m &= \mu_1 \vec{C}_1 + \mu_2 \vec{C}_2 & |\vec{C}_r^*| &= |\vec{C}_r|
 \end{aligned}
 \tag{3-2}$$

The only free parameter in the expressions for post-collision velocities is the orientation of the post-collision relative velocity \vec{C}_r^* . This orientation is uniformly likely to be in any direction in space when hard-spheres collision is assumed [8], as represented by the spherical scattering envelope in

Fig. 3-3. The probability of obtaining \vec{C}_1^* in solid angle element $\Delta^3\vec{\Omega}$ about $-\vec{\Omega}$ (Fig. 3-3) is given by :

$$P(-\vec{\Omega}) = (1/4\pi |\mu_2 \vec{C}_r|^2) (\Delta A / \Delta^3\vec{\Omega}) = (1/4\pi |\cos\delta|) (|\vec{C}_1^*|^2 / |\mu_2 \vec{C}_r|^2) \quad (3-3)$$

where ΔA is an area element on the scattering envelope, whose projection on a plane normal to $\vec{\Omega}$ is $\Delta A |\cos\delta|$. We note that the origin of \vec{C}_m in Fig. 3-3 is external to the scattering envelope, resulting in two possible scattering elements on the sphere. In all the cases that we computed, however (see Ch. 4 below), that point was found to be always internal, so that there was only a single scattering solution with post-collision velocity $\vec{U}_{ik}(S)$ pointing at the spacecraft for any ik pair collision.

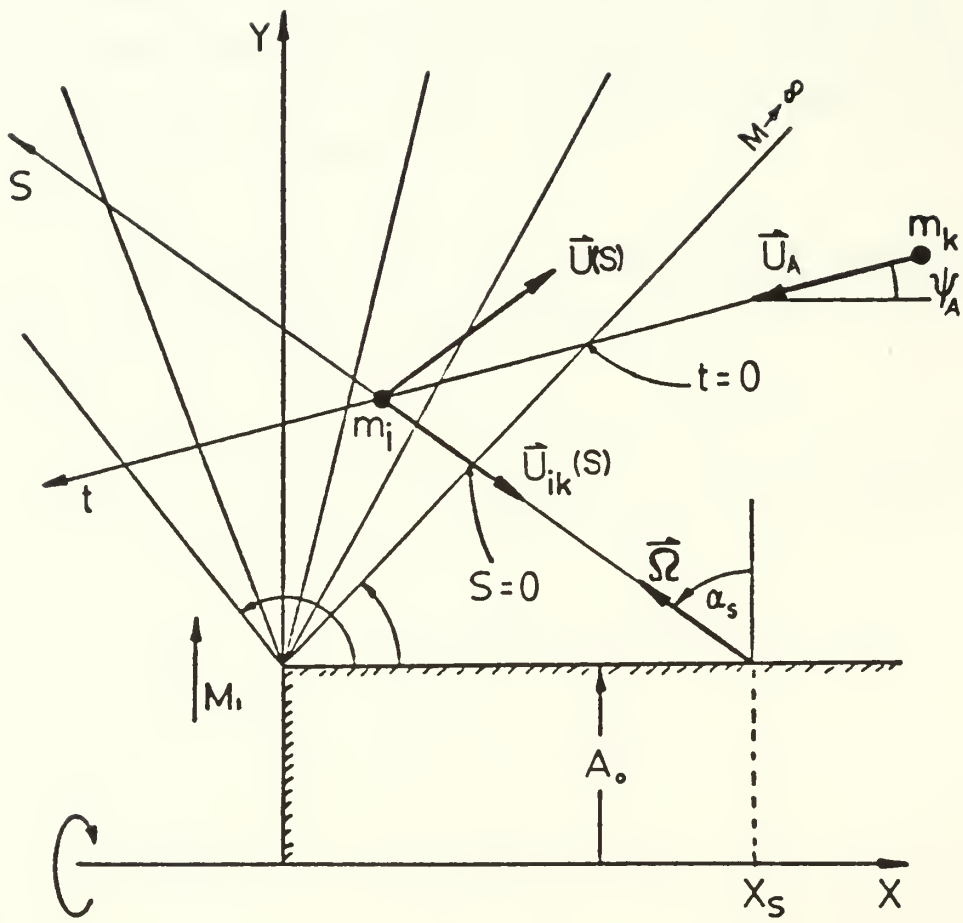


Figure 3-1. Incidence-Plane Description of Flux Integration Scheme.

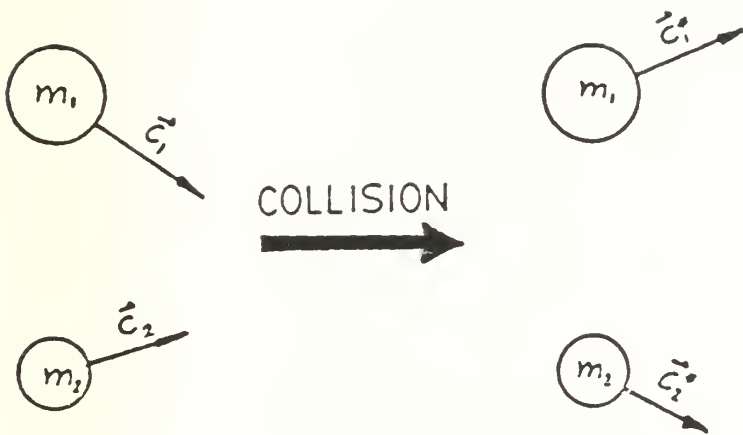


Figure 3-2. Hard-Spheres Collision Notation.

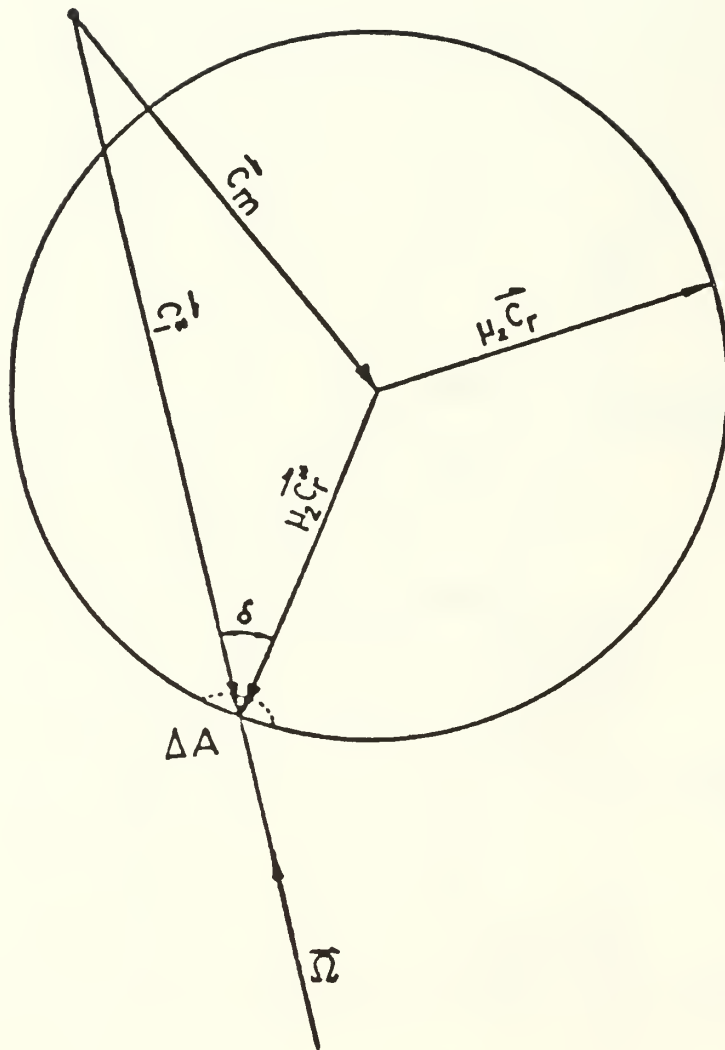


Figure 3-3. Scattering Envelope for Hard-Spheres Collision.

4. RESULTS AND DISCUSSION

We performed several computations of return flux generated by ambient scattering, aimed at demonstrating the expected flux level and its variation with spacecraft target point and orbital attitude angles. In all these computations we assumed that the exhaust flow is as in the typical HF/DF laser case (Table 4-1 below), and that the ambient density and velocity are $N_A = 1 \times 10^{16}$ (molecules/m³) and $U_A = 8$ (km/sec). As an approximation we further assumed that the sole ambient species is molecular nitrogen (molecular weight $W_A = 28$) and that all binary collision cross-sections are uniformly given by $\sigma = \pi D^2$, where D is the molecular diameter (Table 4-1). In each computation we evaluated the combined HF + DF flux by assuming that the molar fraction of DF is zero and the molar fraction of HF is the combined value for both species (Table 4-1): $.091 + .135 = .226$. This is justified by the relatively small difference in molecular weight (just 5%) between these two species.

Three sets of flux computation were performed as follows :

- (a) Incidence-plane ($\phi_A = 0$) target points at various distances from the nozzle lip ($X_s = .1$ to $X_s = 10$ (m)), and at constant incidence angle ($\psi_A = 20^\circ$). The results are shown in Fig. 4-1. We observe that the flux is fairly insensitive to X_s . Also shown in Fig. 4-1 are flux computations where the ring-symmetric CRW flow is approximated as a planar CRW (Prandtl-Meyer flow), rather than the power-law as in Eq. (2-1) above. The planar case exhibits a somewhat higher flux, particularly at large X_s .
- (b) Incidence-plane ($\phi_A = 0$) target points at $X_s = 1$ (m) and at various incidence angles ($\psi_A = 0$ to $\psi_A = 40^\circ$). A polar representation of the results is given in Fig. 4-2. Note the sharp decrease in flux as the incidence angle ψ_A approaches the plume limiting angle $\psi_f = 41^\circ$.
- (c) Azimuth angle variation ($\phi_A = 0$ to $\phi_A = 180^\circ$) at a constant location ($X_s = 1$ (m)) and at a constant angle of incidence ($\psi_A = 20^\circ$). A polar representation of the results is given in Fig. 4-3. Observe that flux becomes sensitive to azimuth angle ϕ_A only past $\phi_A = 90^\circ$, where shadowing by the cylindrical spacecraft becomes increasingly dominant.

In addition to return flux we also computed the rms velocity of the arriving molecules. For the target points in group (a), the rms velocity varied between 6000 and 6600 (m/sec) (the higher velocity at smaller X_s), which corresponds to a kinetic energy of about 4 (ev) per molecule (HF).

The maximum return flux arriving at the spacecraft is about 0.15×10^{19} (molecules/m²sec), which corresponds to a surface deposition rate of about 300 monolayers (HF + DF) per hour. This level of contaminating flux may seem to be not outright negligible; however, since return flux is proportional to ambient density, it will be scaled down considerably at higher altitudes (and lower ambient densities).

We observe that the maximum return flux constitutes a fraction of about 2% of the incident ambient flux. This return flux ratio is roughly maintained at almost all target points and attitude angles in groups (a), (b) and (c). The only exceptions are incidence angles near the limiting cone ($\psi = \psi_r$) or at azimuth angles $\varphi_A > 125^\circ$ where shadowing becomes dominant. This observation is interpreted as follows.

Consider the total solid angle subtended by the limiting cone (considered to be infinitely extended in the axial direction) as viewed from a target point (for all lines-of-sight $\vec{\Omega}$ pointing outward of the cylindrical spacecraft surface). It is independent of target location due to the "self-similar" geometry. During each flux computation, we also evaluated the total solid angle subtended by that segment of the cone over which the flux integration was actually performed (see Section 3.2). It was found out that for all but the "shadowed" cases ($\varphi_A > 125^\circ$), this solid angle constituted a fraction of $86 \pm 1\%$ of the solid angle subtended by the infinite cone. We interpret this result as a hint that geometrical "view factors" arising in the course of the flux integration, are not the dominant factor in determining the 2% level of flux ratio. What then are the dominant factors?

For a possible explanation we turn to the flux integration scheme presented in Section 3.2. The flux ratio is obtained upon dividing the integrand in Eq. (3-1) by $N_A U_A$ and setting $h_k = 1$ (since we assume a single species air). The major factors in the flux ratio integrand appear to be the no-collision probabilities $\exp[-\eta_{ik}(\mathbf{S})]$ and $\exp[-\eta_k(\mathbf{S})]$, and the post-collision directional distribution function $P_{ik}(\mathbf{S}, -\vec{\Omega})$. The flux-averaged values of these functions in the group (a) computations were found to be as follows: $P_{ik}(\mathbf{S}, -\vec{\Omega}) = .09$ to $.10$, $\eta_{ik}(\mathbf{S}) = .42$ to $.54$ and $\eta_k(\mathbf{S}) = .35$ to $.47$. The flux-averaged Mach number for group (a) points exhibited a much larger variation: between 30 and 80, with the higher Mach numbers obtained at further target points.

These results are interpreted as follows. The ambient no-collision probability $\exp[-\eta_k(\mathbf{S})]$ is sufficiently close to unity, so that in an order-of-magnitude analysis such as the present one, we may disregard this factor. If the velocity ratio in the $\eta_{ik}(\mathbf{S})$ integral of Eq. (3-1) is assumed to be unity (its average value for group (a) points is about 1.4), then the differential in the flux \mathbf{S} integration becomes

$\sigma N(\mathbf{S})d\mathbf{S} = d\eta_{ik}(\mathbf{S})$. This implies that the flux \mathbf{S} integration results in some average value of the only remaining factor: $h_i P_{ik}(\mathbf{S}, -\vec{\Omega})$. Since the $\vec{\Omega}$ integration introduces a factor of order unity, the order-of-magnitude estimate for the arriving-to-incident flux ratio is $[h_i P_{ik}(\mathbf{S}, -\vec{\Omega})]_{av}$. The value of this estimate is $[h_i P_{ik}(\mathbf{S}, -\vec{\Omega})]_{av} = .226 \times .09 \approx .02$, which is about equal to the actual flux ratio for target points in group (a).

When an exhaust flow and orbital parameters (velocity and attitude) are specified, $P_{ik}(\mathbf{S}, -\vec{\Omega})$ depends on the choice of molecular collision model (we chose hard spheres), while h_i is uniformly constant. The foregoing reasoning thus establishes the collision model as a significant factor in determining ambient scattering flux levels, to the extent that $P_{ik}(\mathbf{S}, -\vec{\Omega})$ is sensitive to the choice of model.

Table 4-1. Typical Operating Conditions of HF/DF Laser Exhaust

Mole fractions	[H] = .091	[HF] = .091	[H ₂] = .104	[DF] = .135	[He] = .579
Average molecular weight	7.14				
Specific heats ratio	1.54				
Stagnation temperature and density	1400 (K) .0075 (kg/m ³)				
Exit Mach number	4.0				
Molecular diameter (hard spheres)	2.5x10 ⁻¹⁰ (m)				
Spacecraft diameter	5.0 (m)				
Nozzle aperture	1.0 (m)				

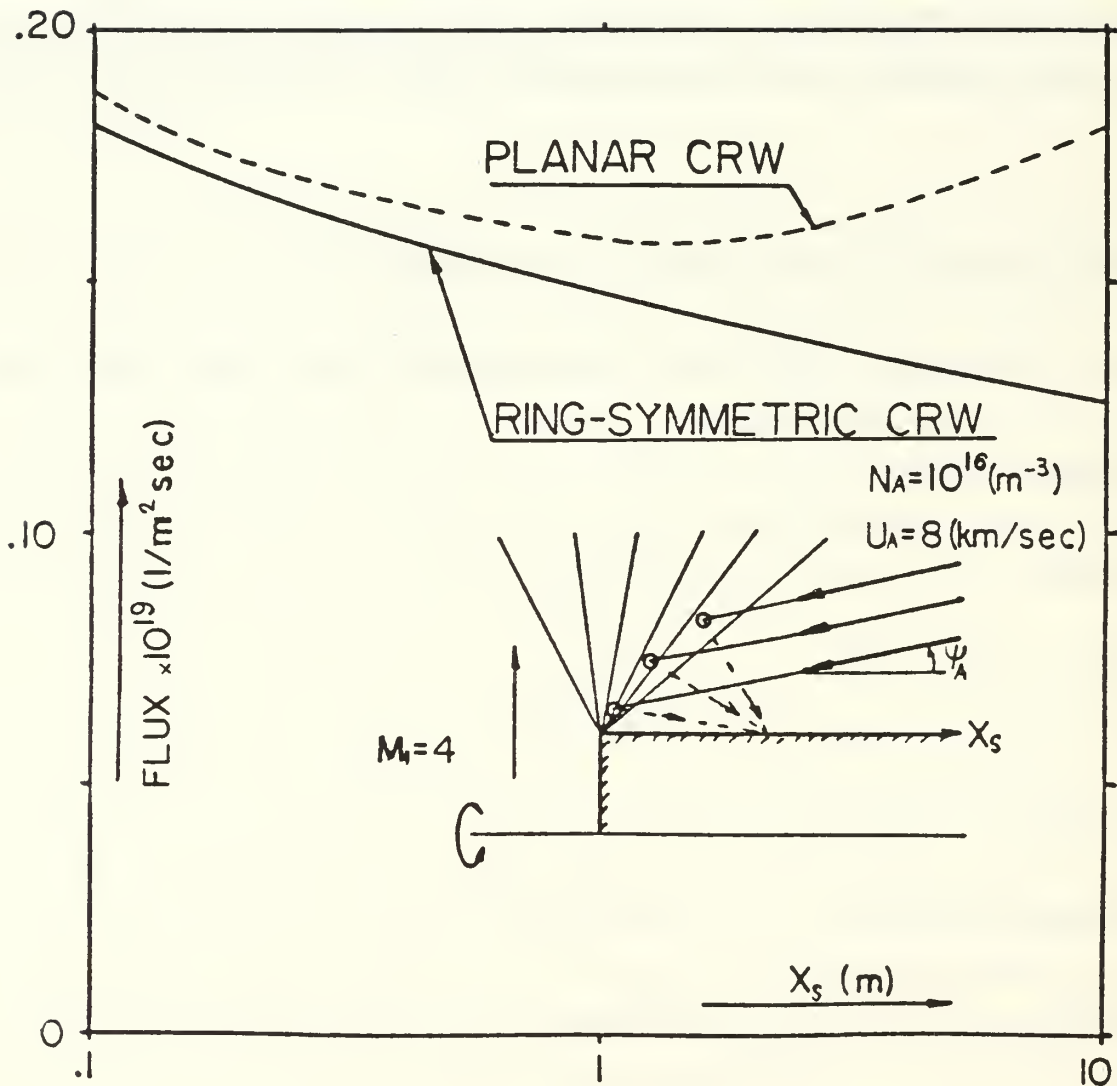


Figure 4-1. Variation of Return Flux with Target Point (X_s). Target Point at Incidence-Plane ($\varphi_A = 0$) and Constant Incidence-Angle ($\psi_A = 20^\circ$).

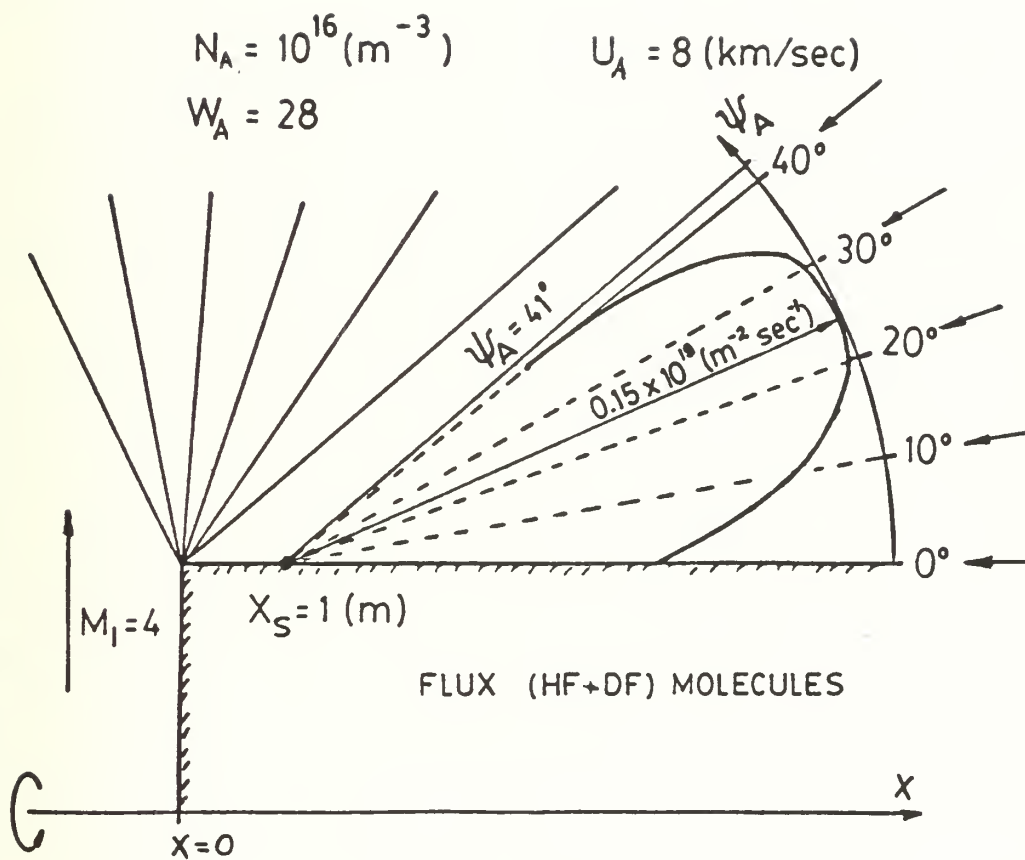


Figure 4-2. Variation of Return Flux with Ambient Incidence Angle (ψ_A). Fixed Target Point ($X_s = 1 \text{ m}$) Located at Incidence-Plane ($\phi_A = 0$).

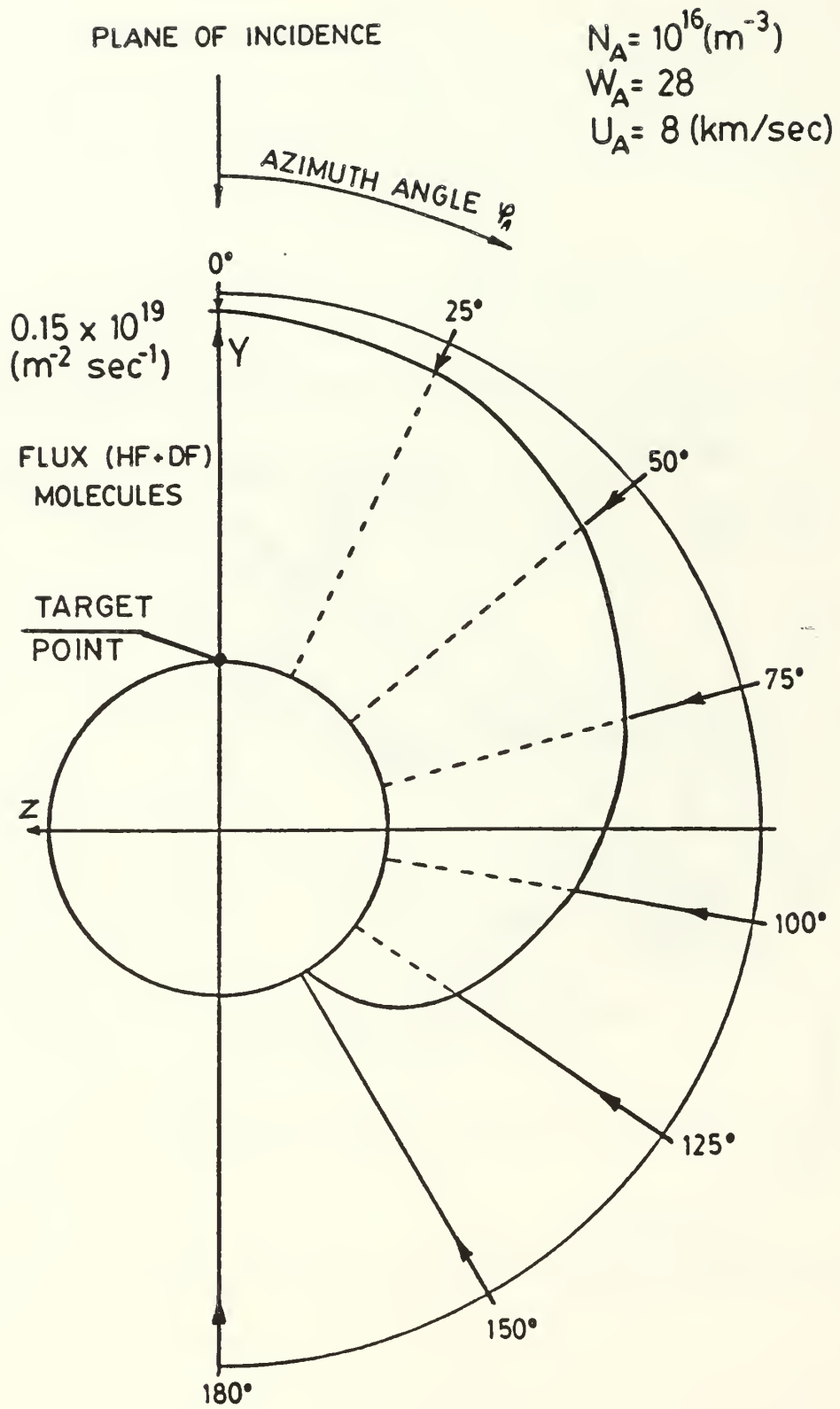


Figure 4-3. Variation of Return Flux with Ambient Azimuth Angle (ψ_A). Fixed Target Point ($X_s = 1 \text{ m}$) and Ambient Incidence Angle ($\psi_A = 20^\circ$).

5. SPACECRAFT CHARGING

Spacecraft charging is a major concern to spacecraft designers, particularly for missions in GEO and to a lesser extent also in LEO. The exhaust plume of an HF/DF laser operating in the ionosphere (300 to 1000 km altitude) may modify significantly the pre-firing charging pattern of the spacecraft. Two classes of effects may lead to charging modification; they are:

- (a) The exhaust contains large concentrations of HF and DF molecules which are highly electronegative. They may be readily ionized by environmental electrons and change the existing spacecraft charging pattern.
- (b) When the spacecraft is oriented obliquely relative to its orbital velocity and the ambient plasma impinges at the plume boundary, the plume will cast a "shadow" on the downstream side, leading to a very dissimilar charging fluxes on the upstream and downstream halves of the spacecraft.

The knowledge gained in analyzing the ambient scattering effect can be applied to the assessment of the effects of ionospheric plasma on spacecraft charging. We first consider the upstream side of the spacecraft as mentioned in (a) above.

We contend that the exhaust-plasma interaction will not drastically alter the charging pattern of the upstream half. This assessment is established as follows. Consider the fact that ionospheric plasma has a particle number density no higher than 10^{12} (m^{-3}) and energy per particle of at most 1 (ev) (excluding the auroral plasma of polar zones or events of sun storms, where the energy per particle is much higher). Significantly, the Debye length at the highest plasma density is very small: only about 10^{-3} (m); the largest Debye length in the ionosphere is 10^{-1} (m) [9]. Ion thermal velocity is typically lower than orbital velocity, but electron velocity is considerably higher than orbital velocity (at 1 ev the electron velocity is about $U_e = 6 \times 10^5$ m/sec). Hence, ions would typically impinge at the plume as a uniform ion beam with the orbital velocity (like ambient molecules), while electrons are expected to impinge at the plume with their random-oriented thermal velocity.

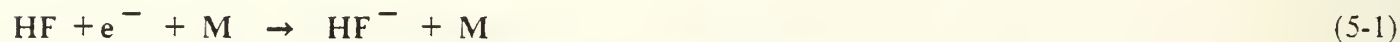
In view of the results of ambient scattering analysis (Ch. 3 and 4 above), and since ions are subject to similar collision process with exhaust molecules as neutrals, ions will be stopped at the plume fringes much like ambient molecules. By virtue of the small Debye length (typically much smaller than the stopping distance), electrons would not penetrate any further than ions, regardless of their

collision cross-section with exhaust molecules. The familiar plasma sheath that forms at a solid surface, is hence replaced at the plume/plasma boundary by a typically neutral layer whose thickness is of the order of an ion/neutral mean free path, but much larger than the Debye length. Only at the upper altitude range of the ionosphere does the Debye length become comparable to a plume boundary mean free path (about .1 m), but there plasma density and flux are several orders of magnitude lower and charging modification is not likely to be significant at the relatively short firing duration of about 5 minutes.

Elastically scattered ions can be deflected towards the spacecraft as a result of elastic collisions with exhaust molecules, much like neutrals. Referring to our analysis of the return-to-ambient flux ratio (Ch. 4 above), it is clear that the relevant ratio here will be about $1/4\pi$, i.e., of the order of 10% (this is due primarily to the role played by the elastic directional distribution function — see Ch. 4). A change in the plasma-to-surface current of that order is hence possible, but unlikely to affect spacecraft design or operation significantly. The reason is that a design capable of smoothing away the inhomogeneous charge flux at oblique attitudes, will not be sensitive to a change in flux pattern of the order of 10% (in other words, potential differences may be amplified by 10%, which is hardly likely in a sound design to bring about arcing or other threshold phenomena).

Another effect which may potentially be significant in the upstream half is generation of electronegative species (HF^- , DF^-) by plasma electrons impinging at the plume. In the sequel, we examine the magnitude of this effect, concluding that it is negligible.

This estimate is best done by considering \dot{N}^- , which is the rate of production of HF^- and DF^- per unit volume, at a typical point in the exhaust where local Mach number is $M=30$ (this is typically the lowest average Mach number for the plume region where ambient scattering takes place — see Ch. 4 above). Since energy is released by the electronegative ion formation, the reaction involves a third body as follows :



where M is the third body molecule. We assume a simplified classical kinetic model for this reaction, as follows. The pair HF/M collide with a frequency proportional to the local number density and HF molar fraction, and to the average relative velocity. An electronegative ion formation can occur only if an electron collides with the pair during their collision, which lasts $t_c = D/\bar{C}_r$, where \bar{C}_r is the average relative pair velocity. Based on this classical model, and assuming the same cross-section for

electronegative ion formation as for elastic HF/M collisions, the volume rate of electronegative ion generation is given by :

$$\dot{N}^- = (\pi D^3 N) N h (\pi D^2 U_e N_e) \quad (5-2)$$

where $(\pi D^3 N)$ is the probability that a certain HF or DF molecule will be in contact (D being molecular hard-sphere diameter) with any other exhaust molecule (whose number density is N). When $(\pi D^3 N)$ is multiplied by hN , where h is the HF + DF molar fraction (Table 4-1), the combined term reads as the number of colliding HF/M pairs per unit volume. Assuming the electronegative formation cross-section is also πD^2 , the factor $\pi D^2 U_e N_e$ where U_e and N_e are electron velocity and number density, renders the expression for electronegative generation rate per unit volume. We note that \bar{C}_r cancels out in deriving Eq. (5-2), so that \dot{N}^- does not depend on temperature. This supports the use of the kinetic approximation in regions of continuum breakdown (plume fringes are such regions).

How is the relative magnitude of \dot{N}^- decided? To do that we multiply \dot{N}^- by $\lambda = 1/\pi D^2 N$, which is the mean free path for a fast moving particle that penetrates the plume. This expression is justified by the fact that most incident particles do collide within a distance of order λ , and when the particles are plasma ions, electrons will adhere to ion spatial distribution by virtue of the small Debye length (smaller than λ). Thus, $\lambda \dot{N}^-$ is the rate of electronegative ion generation per unit area of plume boundary. The ratio β^- between this rate and the incident electron flux is :

$$\beta^- = \lambda \dot{N}^- / N_e U_e = (\pi D^3 N) h = 2.2 \times 10^{-10} \quad (5-3)$$

where $N = 2 \times 10^{19} \text{ (m}^{-3}\text{)}$ which corresponds to Mach number $M = 30$ in the typical case (Table 4-1). The fraction of electron flux captured by HF and DF exhaust molecules to form electronegative ions is so small (due to the pair-formation term $(\pi D^3 N)$), that it cannot appreciably alter the charging flux distribution at the spacecraft surface.

Another possible effect is the recoil of HF^- or DF^- that occurs due to energy released in the electronegative formation reaction. The recoiling species might conceivably reach the surface and contaminate it. The magnitude of the recoil flux is certainly no larger than $\beta^- U_e N_e = 1300 \text{ (m}^{-2} \text{ sec}^{-1}\text{)}$, where we assume the worst case flux : $N_e = 10^{12}$, $U_e = 6 \times 10^5 \text{ (m/sec)}$ which corresponds to about 1 ev energy per electron. This flux level is about 3×10^{-13} monolayers of HF^- and DF^- per hour, so that its contribution to surface contamination is utterly negligible.

The second kind of charging effects (item (b) above) is due to the fact that the exhaust plume is impenetrable to ambient plasma (within a range of sufficiently small distance from the spacecraft, so that no extensive diluting of the plume has taken place). The downstream half of the spacecraft in oblique attitude will be in the "shadow" with respect to incident plasma. As a first approximation we may assume zero plasma flux at the shadowed surface. More accurately, this portion of the spacecraft will be subject to a plasma wake flow. However, it is quite difficult to determine the charging phenomena that take place in such a wake, as indicated by a recent work on solar sails in LEO [9]. Thus, a zero flux at the downstream half seems a practical design assumption.

Can adverse charging effects occur as a result of shadowing the downstream half? This question can be discussed only qualitatively. The reason is that a quantitative analysis requires a lumped-circuit model of the spacecraft external surface [10]. Since such a concrete design is not available, we can only discuss this question qualitatively. Obviously, assuming zero flux to the downstream half during the envisioned 5 minutes of laser firing duration, and requiring that no appreciable voltages between the two halves will evolve, leads to the stipulation that the equivalent-circuit Capacitance \times Resistance should be much smaller than the firing duration.

6. CONCLUDING REMARKS

Our major quantitative conclusion is that for the relatively high ambient density assumed ($N_A = 1 \times 10^{16}$ molecules/m³ which represents Sunspot Maximum at about 200 km) and for the typical HF/DF laser exhaust (Table 4-1), the HF+DF flux backscattered by ambient molecules is several hundred monolayers per hour. This flux level may seem as not outright negligible. However, since ambient scattering flux is proportional to ambient density, it will be scaled down considerably at the lower ambient densities of higher orbital altitudes.

The operational scenario for HF/DF laser envisions 4 or 5 minutes total operating time; hence the contamination by ambient scattering may not be serious due to short operating time.

The effects of laser exhaust plume on spacecraft charging in the ionosphere were examined. It was concluded that the rate of electronegative (HF^- and DF^-) production by impinging electrons was negligible; the low rate is a consequence of the assumption that a third body is required to interact simultaneously with the HF/e or DF/e pair. No significant modification of charging pattern is anticipated. However, at oblique orbital attitudes, the downstream half of the spacecraft will be shadowed from the oncoming ambient plasma. This fact has to be reckoned with in designing a ring-symmetric laser spacecraft.

The emphasis in this work was on the method rather than on results. The first-collision model was demonstrated to be simple to implement in a code. It is considerably simpler than the more general and potentially more accurate Monte Carlo methods commonly used for simulating rarefied flows [8]. We found out that the molecular collision model was all important in determining the return flux level, which is hardly surprising for scattering by single collision. For the same reason, the collision model would also be dominant in a Monte Carlo simulation of the ambient scattering process.

If and when a mathematical accuracy of the first-collision approximation is established for hard-spheres, it might be possible to determine a realistic collision model by comparing computed results with measurements.

This accuracy may be established in either of two ways. One way is by comparison with accurate Monte Carlo computations (using hard-spheres collision model). The other is to seek an estimate of the error incurred by considering just first collisions and ignoring all subsequent ones. This might be achieved by accounting for second collisions in an extended first-collision model, provided a simplified

scheme that will obviate the need for increase in the dimensions of the numerical flux integration can be devised. We are currently considering such second-collision approaches.

7. REFERENCES

- [1] Bird, G. A., "Breakdown of Continuum Flow in Free Jets and Rocket Plumes", Proceedings of 12th Symposium on Rarefied Gas Dynamics. In Volume 74, Progress in Aeronautics and Astronautics, Part II, pp.681, Sam S. Fisher, Editor. Published by AIAA, 1981.
- [2] Falcovitz, J., "Analytic and Numerical Computation of Ring-Symmetric Spacecraft Exhaust Plumes", Report NPS72-86-003CR, Dec. 1986, Naval Postgraduate School, Monterey, CA 93943.
- [3] Liepmann, H. W. and Roshko, A., *Elements of Gasdynamics*, John Wiley, New York, 1957.
- [4] Johnson, F. S., *Satellite Environment Handbook*, Second Edition, Stanford University Press, Stanford, California, 1965.
- [5] Mastrup, F., Broadwell, E., Miller, J. and Jacobs, T. A., "Hydrogen Fluoride Laser Technology Study", Technical Report AFWL-TR-72-28, October 1972.
- [6] Falcovitz, J., "Transient Reactive Exhaust Flow from a Ring-Symmetric HF/DF Space Laser", Report NPS72-87-002CR, Mar. 1987, Naval Postgraduate School, Monterey, CA 93943.
- [7] Falcovitz, J., "Numerical Computation of Ring-Symmetric Spacecraft Exhaust Plumes", Report NPS72-87-001CR, Jan. 1987, Naval Postgraduate School, Monterey, CA 93943.
- [8] Bird, G. A., *Molecular Gas Dynamics*, Clarendon Press, Oxford, 1976.
- [9] Hill, J. R., and Whipple, E. C., "Charging of Large Structures in Space with Application to the Solar Sail Spacecraft (survey)", *Journal of Spacecrafts and Rockets*, Vol. 22, pp. 245-253, 1985.

- [10] Garrett, H. B., "Spacecraft Charging : A Review" , pp. 167-226, in *Space Systems and Their Interactions with Earth's Environment*, Henry B. Garrett and Charles P. Pike, editors, Vol. 71 "Progress in Astronautics and Aeronautics", Published by AIAA, 1980.

APPENDIX A. DESCRIPTION OF AMB CODE

A.1 Description of Subroutines

We provide a list of the subroutines in the ambient flux integration code AMB for ring-symmetric cylindrical spacecrafts. Each subroutine is briefly described. Statements are identified by the FORTRAN statement number (columns 1 through 5).

MAIN PROGRAM The 300 loop is intended to enable several (NCASE) reruns with various data in each, all in a single run. Upon calling INIDAT1, parameters depending on data defined in INIDAT are re-computed. The 200 loop is over various XSV(NX) target points. In the 20 loop the flux integration begins: FLUXC is for particle flux and FLXU2C is for the rms of velocity of return flux molecules. All the MAX suffixed parameters denote values at which the integrand had the largest value.

The actual flux integration commences at statement 1 for the summation over strips of constant RF. This summation is terminated when convergence is attained (to within EPSR). The inner loop 2 is over azimuth angle PHI. Note that the target points are generally not in the plane of incidence (PHIA.NE.0), so that no symmetry can be assumed in the PHI integration, and it is performed twice in order to cover the entire range in PHI (IPAR = 1 for PHI.GT.0, IPAR = 2 for PHI.LT.0). The flux integration along the line-of-sight is done by calling FLUX.

INIDAT Initialization of data. There is no input file for this code. INIDAT1 is for parameters computed from the data defined by calling INIDAT.

SOF Stopping routine, called when an error is detected. Here we also trigger a system error by computing DSQRT(-1), in order to obtain a calling sequence printout by the operating system.

FLUX This routine calls SUMT for flux integration of one exhaust species at a time.

LIMIT Here we compute the point of intersection of the line-of-sight with the leading characteristic cone. If they do not intersect, the distance of the intersecting point TLIM is set to a very large number.

- SUMT This is the line-of-sight integration routine. Runge-Kutta scheme is used (even though an explicit integral is computed). Note that ETAK and ETAIK have to be computed through a separate integration at each point of the line-of-sight integration. The integration step DT ($T = S/RF$) is re-adjusted at each integration step. The integration is terminated when convergence is attained (to within EPST).
- FETA Here the integrand for the line-of-sight flux integration is evaluated. The hard-spheres collision model is used to determine the post-collision directional distribution factor PIK. The flux-average of any variable (such as UIK^{**2} in present version), can be computed by summing it multiplied by flux and subsequently dividing by the total arriving flux (see loop 31 in MAIN PROGRAM).
- PATHIK Here the molecular thickness ETAIK of the I exhaust species scattered by the K ambient molecule, is computed by integration along the line-of-sight.
- FT This routine computes the integrand for the ETAIK integration in PATHIK.
- PATHK The analog to PATHIK for K ambient molecule. TAU is the normalized integration variable along the trajectory of the penetrating ambient molecule. Note that SHADOW=.TRUE. when the trajectory passes through the cylindrical spacecraft surface before entering the fan.
- FTAU Computes the integrand for the ETAK integration in PATHK.
- FAN Computes the fan coordinates PSI, XP, YP for a point on the line-of-sight. It is used to determine the Mach number and flow angle from the power-law approximation (see MATCH).
- FANT Computes the fan coordinates PSI, XP, YP for a point on the ambient molecule trajectory.
- HMSET Prepares the vector HMV(I) which is the value of the H(M) integral at a set of Mach number values (equally spaced in inverse Mach number). This vector is used to compute H(M) for an arbitrary M (see HINTER), since this function is needed in the power-law approximation of flow in a ring-symmetric fan. Subsequent routines MFUNC, HINTER, MATCH and AREAF are all used to implement this approximation.

MFUNC Computes the integrand for the $H(M)$ integration in HMSET.

HINTER Computes $H(M)$ for a given M , from $HMV(I)$ by linear interpolation. Note that the interpolation is done with inverse Mach number as the independent variable.

MATCH Here the approximation to the "inverse problem" of finding the Mach number at a single point in the ring-fan is implemented. An iteration scheme is used to determine the fan characteristic passing through the given point [2].

AREAF Mach number is computed from value of area ratio function. Newton-Raphson iterations are used.

A.2 Listing of AMB code

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C$OPTIONS LIST
C AMBIENT. SCATTERING FROM A RING PLUME BY AMBIENT AIR.
IMPLICIT REAL*8(A-H,O-Z)
COMMON /GAMA/G,G1,G2,G3,G4,G5,G6,G7,G8,G9,G10,G11,G12,G13,G14,G15,
1 G16,G17,G18,G19,G20
COMMON /PAR/CO,ENO,EM1,D,SIGMA,TLIM,DRO,ELO,QO,TO,FACT,ALOGF,
1 DPSIO,DTMAX,DETA0,ETALIM,XSI,XSF
COMMON /NPAR/NPHI,IPAR,NP,NR,NX,NXS,NS,NSPEC,NS1,NS2,NTAU0,NETA0,
1 NAMB,NCASE,ICASE,IFAN
COMMON /GEOM/APF,PAI,PAI2,W,SW,CW,BETA,SBETA,CBETA,PSI1,SPSI1,
1 CPSI1,PSIF,SPSIF,CPSIF,TPSIF,AK,SK,CK,A0,RF,XF,YF,ZF,
2 PHISOF,PHIF,SPHIF,CPHIF,DYMIN,RMIN,XS,DIST,X0,Y0,Z0,
3 DY0,DEG,PSIN,ST1,CT1,OMEGX,OMEGY,OMEGZ,XSV(21)
COMMON /EPSIL/EPSETA,EPST,EPSR
COMMON /EXTREM/TEXT(5),ETAEXT(5),ETAKXT(5),PHIEXT(5),
1 PSIEXT(5),EMEXT(5),FEXT(5),WEXT(5),
2 TMAX(5),ETAKMX(5),ETAMAX(5),PSIMAX(5),
3 EMMAX(5),FMAX(5),
4 RFMAX(5),PHIFMX(5),PHIMAX(5),WMAX(5)
COMMON /COUNTS/ICONTC,ICONTT,ICNTOT,ICNTMX,IQTOT(5),ISHAD(5)
COMMON /SPEC/WAV,XC(5),WC(5),WRC(5),XNAME(5),QFC(5),QDC(5),
1 QU2C(5),FLUXC(5),OMEGA(5),FLXU2C(5),URMSC(5)
COMMON /AMBIEN/ENA,UA,PSIA,PHIA,HA(3),WA(3),
1 UAX,UAY,UAZ,AA,BA,CA,RA,XA,YA,ZA,SHADOW
COMMON /POINT/XP,YP,XCOR,YCOR
LOGICAL SHADOW
DIMENSION DSUMF(5),DSUMD(5),DSUMAX(5),DSUMU2(5)
NCASE=1
DO 300 ICASE=1,NCASE
CALL INIDAT
GO TO (301,302,303,304,305,306,307,308,309,310,
1 311,312,313,314,315,316,317,318,319,320),ICASE
301 CONTINUE
IFAN=1
NXS=3
XSI=0.1D0
GO TO 399
302 CONTINUE
PHIA=20.D0/DEG
GO TO 399
303 CONTINUE
PHIA=50.D0/DEG
GO TO 399
304 CONTINUE
PHIA=75.D0/DEG
GO TO 399
305 CONTINUE
PHIA=100.D0/DEG
GO TO 399
306 CONTINUE
PHIA=125.D0/DEG
GO TO 399
307 CONTINUE
PHIA=150.D0/DEG
GO TO 399
308 CONTINUE
PHIA=175.D0/DEG
GO TO 399
309 CONTINUE
GO TO 399
310 CONTINUE
GO TO 399
311 CONTINUE
GO TO 399
312 CONTINUE
GO TO 399
313 CONTINUE
GO TO 399
314 CONTINUE
GO TO 399
315 CONTINUE
GO TO 399

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316 CONTINUE                                AMB0073
GO TO 399                                  AMB0074
317 CONTINUE                                AMB0075
GO TO 399                                  AMB0076
318 CONTINUE                                AMB0077
GO TO 399                                  AMB0078
319 CONTINUE                                AMB0079
GO TO 399                                  AMB0080
320 CONTINUE                                AMB0081
GO TO 399                                  AMB0082
399 CONTINUE                                AMB0083
PRINT 101                                  AMB0084
101 FORMAT('1')                             AMB0085
C                                            AMB0086
CALL INDAT1                                 AMB0087
C                                            AMB0088
DO 200 NX=1,NXS                             AMB0089
XS=XSV(NX)                                  AMB0090
C (X0,Y0,Z0) IS THE POINT AT WHICH FLUX AND DENSITY ARE COMPUTED. AMB0091
C THE NORMAL TO THE SURFACE AT (X0,Y0,Z0) IS PARALLEL TO Y-AXIS. AMB0092
X0=XS                                        AMB0093
Y0=A0                                        AMB0094
Z0=0.                                        AMB0095
DO 20 NS=NS1,NS2                            AMB0096
FLUXC(NS)=0.                                AMB0097
FLXU2C(NS)=0.                               AMB0098
OMEGA(NS)=0.                                 AMB0099
DSUMAX(NS)=0.                               AMB0100
IQTOT(NS)=0                                 AMB0101
ISHAD(NS)=0.                                AMB0102
TMAX(NS)=-1.D 44                            AMB0103
ETAKMX(NS)=-1.D 44                          AMB0104
PHIMAX(NS)=-1.D 44                          AMB0105
PHIFMX(NS)=-1.D 44                          AMB0106
WMAX(NS)=-1.D 44                            AMB0107
PSIMAX(NS)=-1.D 44                          AMB0108
ETAMAX(NS)=-1.D 44                          AMB0109
RFMAX(NS)=-1.D 44                           AMB0110
EMMAX(NS)=-1.D 44                           AMB0111
FMAX(NS)=-1.D 44                            AMB0112
20 CONTINUE                                  AMB0113
RN=RMIN                                      AMB0114
APF=A0-0.5D0*DR0*SPSIF                      AMB0115
NR=0                                         AMB0116
1 NR=NR+1                                    AMB0117
DR=DR0                                       AMB0118
DR=DR0*(APF/A0)                              AMB0119
C DR=DR0*(1.D0+0.4D0*DR0/XS)**NR           AMB0120
RF=RN+DR/2.D0                                AMB0121
APF=A0+RF*SPSIF                              AMB0122
PHISOF=DACOS(A0/APF)                          AMB0123
C DPHIO=0.1D0                                AMB0124
C NPHI=PHISOF/DPHI0+2                         AMB0125
DPHI=PHISOF/DBLE(NPHI)                       AMB0126
DO 21 NS=NS1,NS2                             AMB0127
DSUMF(NS)=0.                                 AMB0128
DSUMU2(NS)=0.                                AMB0129
DSUMD(NS)=0.                                 AMB0130
21 CONTINUE                                  AMB0131
DOMEGR=0.                                     AMB0132
DO 2 NP=1,NPHI                                AMB0133
DO 2 IPAR=1,2                                  AMB0134
PHIF=(DBLE(NP)-0.5D0)*DPHI                   AMB0135
IF(IPAR.EQ.2) PHIF=-PHIF                     AMB0136
C                                            AMB0137
CALL FLUX                                     AMB0138
C                                            AMB0139
CROSS1=OMEGY                                  AMB0140
CROSS2=(SPSIF)*(-OMEGX)+(-CPSIF*CPHIF)*(-OMEGY)+ AMB0141
1 (-CPSIF*SPHIF)*(-OMEGZ)                   AMB0142
IF(CROSS1.LE.0.)                              AMB0143
1CALL SOF('DIRECTION COSINE OF SURFACE NORMAL SHOULD BE POSITIVE') AMB0144

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IF(CROSS2.LE.0.)
1CALL SOF('NORMAL TO LIMITING CONE HAS NEGATIVE PROJECTION ON LINE-
1OF-SIGHT')
DOMEGA=CROSS2*DPHI*APF*DR/DIST**2
DOMEGR=DOMEGR+DOMEGA
DO 24 NS=NS1,NS2
DSUMF(NS)=DSUMF(NS)+DOMEGA*QFC(NS)*CROSS1
DSUMU2(NS)=DSUMU2(NS)+DOMEGA*QU2C(NS)*CROSS1
IF(DSUMAX(NS).GT.DOMEGA*QFC(NS)*CROSS1) GO TO 24
DSUMAX(NS)=DOMEGA*QFC(NS)*CROSS1
TMAX(NS)=TEXT(NS)
ETAKMX(NS)=ETAKXT(NS)
PHIMAX(NS)=PHIEXT(NS)*DEG
PHIFMX(NS)=PHIF*DEG
WMAX(NS)=WEXT(NS)*DEG
PSIMAX(NS)=PSIEXT(NS)*DEG
ETAMAX(NS)=ETAEXT(NS)
RFMAX(NS)=RF
EMMAX(NS)=EMEXT(NS)
FMAX(NS)=QFC(NS)*XC(NS)*Q0
24 CONTINUE
2 CONTINUE
DO 26 NS=NS1,NS2
FLUXC(NS)=FLUXC(NS)+DSUMF(NS)
FLXU2C(NS)=FLXU2C(NS)+DSUMU2(NS)
OMEGA(NS)=OMEGA(NS)+DOMEGR
26 CONTINUE
RN=RN+DR
IF(NR.LE.2) GO TO 1
IF(NR.GT.99) GO TO 10
DO 27 NS=NS1,NS2
IF(FLUXC(NS).EQ.0.) GO TO 27
ERR=(DSUMF(NS)/FLUXC(NS))/DOMEGR
IF(ERR.GT.EPSR) GO TO 28
27 CONTINUE
GO TO 10
28 CONTINUE
GO TO 1
10 CONTINUE
DO 31 NS=NS1,NS2
FLUXC(NS)=XC(NS)*FLUXC(NS)*Q0
OMEGA(NS)=OMEGA(NS)/(2.D0*PAI*DCOS(PSIF/2.D0)**2)
FLXU2C(NS)=XC(NS)*FLXU2C(NS)*Q0
URMSC(NS)=0.
IF(FLUXC(NS).EQ.0.) GO TO 31
URMSC(NS)=DSQRT(FLXU2C(NS)/FLUXC(NS))
C AVERAGE EM (SEE FETA)
URMSC(NS)= FLXU2C(NS)/FLUXC(NS)
31 CONTINUE
PRINT 11,NX,NR,XS,RF,DR,PHISOF*DEG
11 FORMAT(//1X,'NX,NR,XS,RF,DR,PHISOF=',2I4,3D13.4,F8.4,
1 3X,'FLUX AND EXTREMA VALUES, ALL SPECIES:')
PRINT 12
12 FORMAT(/1X,' NAME ',' IQTOT',' ISHAD',
1 ' FMAX ',' OMEGA',' TMAX',
2 ' ETAKMX',' ETAMAX',' PSIMAX',
3 ' EMMAX',' RFMAX',' PI-WMAX',
4 ' URMSC',' FLUXC / LOG'/)
DO 14 NS=NS1,NS2
DLF=0.
IF(FLUXC(NS).NE.0)
1DLF=DLOG10(FLUXC(NS))+100.D0+1.D-11
IDLF=DLF
DLF=DLF-DBLE(IDLF)
PRINT 13,XNAME(NS),IQTOT(NS),ISHAD(NS),FMAX(NS),OMEGA(NS),
1 TMAX(NS),ETAKMX(NS),ETAMAX(NS),
2 PSIMAX(NS),EMMAX(NS),RFMAX(NS),
3 180.D0-WMAX(NS),URMSC(NS),
4 FLUXC(NS),DLF
13 FORMAT(1X,A6,2I6,D10.3,4F8.4,4F8.1,F8.2,D10.3,'/',F4.2)
14 CONTINUE
200 CONTINUE

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AMB0145
 AMB0146
 AMB0147
 AMB0148
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 AMB0168
 AMB0169
 AMB0170
 AMB0171
 AMB0172
 AMB0173
 AMB0174
 AMB0175
 AMB0176
 AMB0177
 AMB0178
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 AMB0192
 AMB0193
 AMB0194
 AMB0195
 AMB0196
 AMB0197
 AMB0198
 AMB0199
 AMB0200
 AMB0201
 AMB0202
 AMB0203
 AMB0204
 AMB0205
 AMB0206
 AMB0207
 AMB0208
 AMB0209
 AMB0210
 AMB0211
 AMB0212
 AMB0213
 AMB0214
 AMB0215
 AMB0216


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102 PRINT 102
300 FORMAT(///1X,'END RING RUN',///)
CONTINUE
STOP
END
SUBROUTINE INIDAT
IMPLICIT REAL*8(A-H,O-Z)
REAL*8 LAMDA0,LAMDA1
CHARACTER*8 XNAME,XNAMED
COMMON /GAMA/G,G1,G2,G3,G4,G5,G6,G7,G8,G9,G10,G11,G12,G13,G14,G15,
1 G16,G17,G18,G19,G20
COMMON /PAR/CO,ENO,EM1,D,SIGMA,TLIM,DR0,EL0,Q0,T0,FACT,ALOGF,
1 DPSI0,DTMAX,DETA0,ETALIM,XSI,XSF
COMMON /NPAR/NPHI,IPAR,NP,NR,NX,NXS,NS,NSPEC,NS1,NS2,NTAU0,NETA0,
1 NAMB,NCASE,ICASE,IFAN
COMMON /GEOM/APF,PAI,PAI2,W,SW,CW,BETA,SBETA,CBETA,PSI1,SPSI1,
1 CPSI1,PSIF,SPSIF,CPSIF,TPSIF,AK,SK,CK,A0,RF,XF,YF,ZF,
2 PHISOF,PHIF,SPHIF,CPHIF,DYMIN,RMIN,XS,DIST,X0,Y0,Z0,
3 DY0,DEG,PSIN,ST1,CT1,OMEGX,OMEGY,OMEGZ,XSV(21)
COMMON /EPSIL/EPSETA,EPST,EPSR
COMMON /EXTREM/TEXT(5),ETAEXT(5),ETAKXT(5),PHIEXT(5),
1 PSIEXT(5),EMEXT(5),FEXT(5),WEXT(5),
2 TMAX(5),ETAKMX(5),ETAMAX(5),PSIMAX(5),
3 EMMAX(5),FMAX(5),
4 RFMAX(5),PHIFMX(5),PHIMAX(5),WMAX(5)
COMMON /COUNTS/ICONTC,ICONTT,ICNTOT,ICNTMX,IQTOT(5),ISHAD(5)
COMMON /SPEC/WAV,XC(5),WC(5),WRC(5),XNAME(5),QFC(5),QDC(5),
1 QU2C(5),FLUXC(5),OMEGA(5),FLXU2C(5),URMSC(5)
COMMON /AMBIEN/ENA,UA,PSIA,PHIA,HA(3),WA(3),
1 UAX,UAY,UAZ,AA,BA,CA,RA,XA,YA,ZA,SHADOW
COMMON /POINT/XP,YP,XCOR,YCOR
LOGICAL SHADOW
DIMENSION XCD(5),WCD(5),XNAMED(5)
DATA XCD/.091D0,.091D0,.104D0,.135D0,.579D0/
DATA WCD/1.00D0,20.0D0,2.00D0,21.0D0,4.00D0/
DATA XNAMED/' H ',' HF ',' H2 ',' DF ',' HE '/'
DATA IFIRST/0/
IFAN=2
PAI=4.D0*DATAN(1.D0)
PAI2=PAI/2.D0
DEG=180.D0/PAI
AR=8.3143D3
AV=6.022D 26
C OMEGAC=0.5 IS FOR HARD SPHERE COLLISIONS,
C AN AVERAGE RECOMMENDED VALUE IS ABOUT OMEGAC=0.75
OMEGAC=0.5D0
NSPEC=5
NS1=2
NS2=2
DO 51 NS=1,NSPEC
XC(NS)=XCD(NS)
WC(NS)=WCD(NS)
XNAME(NS)=XNAMED(NS)
51 CONTINUE
C COMBINE HF AND DF MOLE FRACTIONS INTO HF FRACTION
XC(2)=XC(2)+XC(4)
XC(4)=0.
C
A0=2.5D0
EM1=4.D0
RH00=0.0075D0
T0=1400.D0
G=1.54D0
D=2.5D-10
NXS=1
XSI=1.0D0
XSF=10.D0
C AMBIENT AIR
ENA=1.00D 16
UA=8.D 3
NAMB=3
WA(1)=28.D0

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WA(2)=32.D0
WA(3)=16.D0
HA(1)=1.D0
HA(2)=0.
HA(3)=0.
PSIA=20.D0/DEG
PHIA=0.00D0/DEG
C INTEGRATION PARAMETERS
NPHI=6
NTAU0=4
NETA0=4
ICNTMX=100
RMIN=0.
DRO=0.10D0
DPSI0=0.20D0
DTMAX=1.0D0
DETA0=0.50D0
ETALIM=10.D0
EPST=0.5D0
EPSR=0.3D0
FACT=1.D 20
RETURN
C*****
C COMPUTATION OF DATA-DEPENDENT PARAMETERS
C*****
ENTRY INDAT1
C*****
ALOGF=DLOG(FACT)
WAV=0.
DO 52 NS=1, NSPEC
WAV=WAV+XC(NS)*WC(NS)
52 CONTINUE
DO 53 NS=1, NSPEC
WRC(NS)=WC(NS)/WAV
53 CONTINUE
SIGMA=PAI*D**2
EN0=RHO0*AV/WAV
C0=DSQRT(G*AR*T0/WAV)
XSV(1)=XSI
IF(NXS.EQ.1) GO TO 12
DXL=(DLOG(XSF)-DLOG(XSI))/(DBLE(NXS)-1.D0)
XLI=DLOG(XSI)
DO 11 NX=2, NXS
XSV(NX)=DEXP(XLI+(DBLE(NX)-1.D0)*DXL)
11 CONTINUE
12 CONTINUE
G1=(G-1.D0)/2.D0
G2=(G+1.D0)/(2.D0*(G-1.D0))
G3=G/2.D0
G4=(G+1.D0)/(G-1.D0)
G5=DSQRT((G+1.D0)/(G-1.D0))
G6=1.D0/(G-1.D0)
G7=2.D0/(G+1.D0)
G8=(0.5D0*(G+1.D0)**2/(G-1.D0))*((1.D0/(G+1.D0))*
1 ((G+1.D0)/(G-1.D0))*((G-1.D0)/(G+1.D0))
G9=(G+3.D0)/(2.D0*(G-1.D0))
G10=(7.D0-3.D0*G)/(2.D0*(G-1.D0))
G11=(5.D0-3.D0*G)/(2.D0*(G-1.D0))
G13=(2.D0-G)/(2.D0*(G-1.D0))
G14=G/(2.D0*(G-1.D0))
G15=(G+1.D0)/(3.D0-G)
ZETA1=G5*DATAN(DSQRT(EM1**2-1.D0)/G5)
AMU1=DASIN(1.D0/EM1)
PSI1=PAI2+AMU1
SPSI1=DSIN(PSI1)
CPSI1=DCOS(PSI1)
PSIF=PAI2+AMU1+ZETA1-G5*PAI2
SPSIF=DSIN(PSIF)
CPSIF=DCOS(PSIF)
TPSIF=DTAN(PSIF)
TETA1=PSI1-AMU1
ST1=DSIN(TETA1)

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AMB0289
AMB0290
AMB0291
AMB0292
AMB0293
AMB0294
AMB0295
AMB0296
AMB0297
AMB0298
AMB0299
AMB0300
AMB0301
AMB0302
AMB0303
AMB0304
AMB0305
AMB0306
AMB0307
AMB0308
AMB0309
AMB0310
AMB0311
AMB0312
AMB0313
AMB0314
AMB0315
AMB0316
AMB0317
AMB0318
AMB0319
AMB0320
AMB0321
AMB0322
AMB0323
AMB0324
AMB0325
AMB0326
AMB0327
AMB0328
AMB0329
AMB0330
AMB0331
AMB0332
AMB0333
AMB0334
AMB0335
AMB0336
AMB0337
AMB0338
AMB0339
AMB0340
AMB0341
AMB0342
AMB0343
AMB0344
AMB0345
AMB0346
AMB0347
AMB0348
AMB0349
AMB0350
AMB0351
AMB0352
AMB0353
AMB0354
AMB0355
AMB0356
AMB0357
AMB0358
AMB0359
AMB0360

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CT1=DCOS(TETA1)
Q0=ENA*UA
LAMDA0=1.D0/(DSQRT(2.D0)*SIGMA*ENO)
LAMDA1=LAMDA0*(1.D0+G1*EM1**2)**(G6-OMEGAC+0.5D0)
AA=DCOS(PSIA)
BA=DSIN(PSIA)*DCOS(PHIA)
CA=DSIN(PSIA)*DSIN(PHIA)
UAX=-UA*AA
UAY=-UA*BA
UAZ=-UA*CA
XCOR=0.
YCOR=A0
C
PRINT 201, NSPEC, XNAME
201 FORMAT(/1X, 'SPECIES DATA NSPEC=', I3/
1 1X, 'SPECIES NAMES ', 11(2X, A6, 2X))
PRINT 202, XC
202 FORMAT( 1X, 'MOLE FRACTION XC=', 11(F8.4, 2X))
PRINT 203, WC
203 FORMAT( 1X, 'MOL. WEIGHT WC=', 11(F8.4, 2X))
PRINT 21, AR, AV, WAV, G, RH00, T0, EN0, C0, D
21 FORMAT(/1X, 'THERMODYNAMIC DATA'/
1 1X, 'AR, AV, WAV, GAMMA=', 2X, 2D14.5, 2F9.3/
2 1X, 'RH00, T0, EN0, C0, D=', D12.4, F8.0, D13.5, 2D12.4)
PRINT 22, EM1, PS11*DEG, PSIF*DEG,
1 A0, LAMDA0, LAMDA1
22 FORMAT(/1X, 'FLOW AND GEOMETRY DATA'/
1 1X, 'EM1, PS11, PSIF=', 3F9.3/
2 1X, 'A0, LAMDA0, LAMDA1=', F9.3, 2D13.4)
PRINT 23, DPSI0, DTMAX, DETA0, ETALIM, DR0, RMIN,
1 EPST, EPSR,
2 NPHI, NTAU0, NETA0
23 FORMAT(/1X, 'INTEGRATION DATA'/
1 1X, 'DPSI0, DTMAX, DETA0, ETALIM=', 4F9.4/
2 1X, 'DR0, RMIN, =', 2D13.4/
3 1X, 'EPST, EPSR=', 2D12.3/
4 1X, 'NPHI, NTAU0, NETA0=', 3I6)
PRINT 24, ENA, UA, PSIA*DEG, PHIA*DEG
24 FORMAT(/1X, 'ABBREVIATED AIR DATA'/
1 1X, 'ENA, UA=', 2D13.4/
2 1X, 'PSIA, PHIA=', 2F9.1)
GO TO (251, 252), IFAN

251 CONTINUE
PRINT 2510, IFAN
2510 FORMAT(/1X, 'RING-FAN APPROXIMATED AS PLANAR. IFAN=', I4)
GO TO 250
252 CONTINUE
PRINT 2520, IFAN
2520 FORMAT(/1X, 'RING-FAN APPROXIMATED BY MATCHED APPROXIMATION.',
1 4X, 'IFAN=', I4)
250 CONTINUE
PRINT 29
29 FORMAT(/1X, 'END DATA'///)
IF(IFIRST.EQ.0.AND.IFAN.EQ.2)
1CALL HMSET
IF(IFAN.EQ.2) IFIRST=IFIRST+1
RETURN
END
C$OPTIONS LIST
SUBROUTINE SOF(ISTOP)
IMPLICIT REAL*8(A-H, O-Z)
CHARACTER*4 ISTOP(1)
COMMON /GAMA/G, G1, G2, G3, G4, G5, G6, G7, G8, G9, G10, G11, G12, G13, G14, G15,
1 G16, G17, G18, G19, G20
COMMON /PAR/C0, EN0, EM1, D, SIGMA, TLIM, DR0, EL0, Q0, T0, FACT, ALOGF,
1 DPSI0, DTMAX, DETA0, ETALIM, XSI, XSF
COMMON /NPAR/NPHI, IPAR, NP, NR, NX, NXS, NS, NSPEC, NS1, NS2, NTAU0, NETA0,
1 NAMB, NCASE, ICASE, IFAN
COMMON /GEOM/APF, PAI, PAI2, W, SW, CW, BETA, SBETA, CBETA, PS11, SPS11,
1 CPS11, PSIF, SPSIF, CPSIF, TPSIF, AK, SK, CK, A0, RF, XF, YF, ZF,
2 PHISOF, PHIF, SPHIF, CPHIF, DYMIN, RMIN, XS, DIST, X0, Y0, Z0,

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3          DY0, DEG, PSIN, ST1, CT1, OMEGX, OMEGY, OMEGZ, XSV(21)          AMB0433
COMMON /EPSIL/EPSETA, EPST, EPSR                                          AMB0434
COMMON /EXTREM/TEXT(5), ETAEXT(5), ETAKXT(5), PHIEXT(5),                 AMB0435
1          PSIEXT(5), EMEXT(5), FEXT(5), WEXT(5),                          AMB0436
2          TMAX(5), ETAKMX(5), ETAMAX(5), PSIMAX(5),                      AMB0437
3          EMMAX(5), FMAX(5),                                             AMB0438
4          RFMAX(5), PHIFMX(5), PHIMAX(5), WMAX(5)                        AMB0439
COMMON /SOFPR/C, DSUMF, DSUMD, T, ETA, DETA, SUM, DSUM, SUMU, DSUMU      AMB0440
COMMON /SUMS/SUMF(5), SUMD(5), SUMU2(5)                                    AMB0441
COMMON /COUNTS/ICONTC, ICONTT, ICNTOT, ICNTMX, IQTOT(5), ISHAD(5)      AMB0442
COMMON /SPEC/WAV, XC(5), WC(5), WRC(5), XNAME(5), QFC(5), QDC(5),       AMB0443
1          QU2C(5), FLUXC(5), OMEGA(5), FLXU2C(5), URMSC(5)             AMB0444
PRINT 1, ISTOP                                                            AMB0445
1  FORMAT(///1X, 2H**, 2X, 30A4, 2X, 2H**, ///)                            AMB0446
PRINT 71, NS, NP, NR, NX, ICONTC, ICONTT                                  AMB0447
71  FORMAT(1X, 'NS, NP, NR, NX, ICONTC, ICONTT=', 6I6/)                  AMB0448
IF(NS.GT.NSPEC) NS=1                                                     AMB0449
PRINT 72, RF, PHIF*DEG, PHISOF*DEG, W*DEG, BETA*DEG                     AMB0450
72  FORMAT(/1X, 'RF, PHIF, PHISOF, W, BETA=', D14.5, 4F10.3/)          AMB0451
PRINT 73, C, T, TLIM, ETA                                               AMB0452
73  FORMAT(/1X, 'C, T, TLIM, ETA=', 4D14.5/)                            AMB0453
PRINT 74, DSUM, SUM, DSUMF, SUMF(NS), SUMD(NS), QDC(NS), QFC(NS),       AMB0454
1          FLUXC(NS), OMEGA(NS)                                          AMB0455
74  FORMAT(1X, 'DSUM, SUM, DSUMF, SUMF(NS), SUMD(NS)=', 5D14.5/        AMB0456
1          1X, 'QDC(NS), QFC(NS), FLUXC(NS), OMEGA(NS)=', 4D14.5/)     AMB0457
XX=-1.DO                                                                  AMB0458
YY=DSQRT(XX)+1.DO                                                         AMB0459
STOP                                                                        AMB0460
END                                                                          AMB0461
SUBROUTINE FLUX                                                            AMB0462
IMPLICIT REAL*8(A-H, O-Z)                                                AMB0463
COMMON /GAMA/G, G1, G2, G3, G4, G5, G6, G7, G8, G9, G10, G11, G12, G13, G14, G15, AMB0464
1          G16, G17, G18, G19, G20                                       AMB0465
COMMON /PAR/CO, EN0, EM1, D, SIGMA, TLIM, DR0, EL0, Q0, T0, FACT, ALOGF, AMB0466
1          DPSI0, DTMAX, DETA0, ETALIM, XSI, XSF                          AMB0467
COMMON /NPAR/NPHI, IPAR, NP, NR, NX, NXS, NS, NSPEC, NS1, NS2, NTAU0, NETA0, AMB0468
1          NAMB, NCASE, ICASE, IFAN                                       AMB0469
COMMON /GEOM/APF, PAI, PAI2, W, SW, CW, BETA, SBETA, CBETA, PSI1, SPSI1, AMB0470
1          CPSI1, PSIF, SPSIF, CPSIF, TPSIF, AK, SK, CK, A0, RF, XF, YF, ZF, AMB0471
2          PHISOF, PHIF, SPHIF, CPHIF, DYMIN, RMIN, XS, DIST, X0, Y0, Z0, AMB0472
3          DY0, DEG, PSIN, ST1, CT1, OMEGX, OMEGY, OMEGZ, XSV(21)       AMB0473
COMMON /EPSIL/EPSETA, EPST, EPSR                                          AMB0474
COMMON /EXTREM/TEXT(5), ETAEXT(5), ETAKXT(5), PHIEXT(5),                 AMB0475
1          PSIEXT(5), EMEXT(5), FEXT(5), WEXT(5),                          AMB0476
2          TMAX(5), ETAKMX(5), ETAMAX(5), PSIMAX(5),                      AMB0477
3          EMMAX(5), FMAX(5),                                             AMB0478
4          RFMAX(5), PHIFMX(5), PHIMAX(5), WMAX(5)                        AMB0479
COMMON /SOFPR/C, DSUMF, DSUMD, T, ETA, DETA, SUM, DSUM, SUMU, DSUMU      AMB0480
COMMON /COUNTS/ICONTC, ICONTT, ICNTOT, ICNTMX, IQTOT(5), ISHAD(5)      AMB0481
COMMON /SPEC/WAV, XC(5), WC(5), WRC(5), XNAME(5), QFC(5), QDC(5),       AMB0482
1          QU2C(5), FLUXC(5), OMEGA(5), FLXU2C(5), URMSC(5)             AMB0483
COMMON /SUMS/SUMF(5), SUMD(5), SUMU2(5)                                    AMB0484
EL0=SIGMA*RF*ENO                                                         AMB0485
IF(Z0.NE.0.)                                                              AMB0486
1CALL SOF('THE SCHEME HERE IS NOT WRITTEN FOR Z0.NE.0.')
```

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W=PAI2-DATAN(CW/SW)
OMEGX=CW
OMEGY=SW*CBETA
OMEGZ=SW*SBETA
CALL LIMIT
C
DO 20 NS=NS1,NS2
SUMF(NS)=0.
SUMU2(NS)=0.
SUMD(NS)=0.
FEXT(NS)=0.
CALL SUMT
SUMF(NS)=SUM
SUMU2(NS)=SUMU
QFC(NS)=SUMF(NS)/FACT
QU2C(NS)=SUMU2(NS)/FACT
FEXT(NS)=FEXT(NS)/FACT
CALL FAN(TEXT(NS),PSIEXT(NS),PHIEXT(NS))
IF(PSIEXT(NS).LT.PSIF-1.D-10) CALL SOF('PSIEXT(NS).LT.PSIF')
IF(PSIEXT(NS).GT.PSI1) PSIEXT(NS)=PSI1
PSI0=PSIEXT(NS)
T=TEXT(NS)
CALL MATCH(T,PSI0,EM,TETA)
EMEXT(NS)=EM
WEXT(NS)=W
IQTOT(NS)=IQTOT(NS)+ICONTT
20 CONTINUE
RETURN
END
SUBROUTINE LIMIT
IMPLICIT REAL*8(A-H,O-Z)
COMMON /GAMA/G,G1,G2,G3,G4,G5,G6,G7,G8,G9,G10,G11,G12,G13,G14,G15,
1 G16,G17,G18,G19,G20
COMMON /PAR/CO,ENO,EM1,D,SIGMA,TLIM,DR0,EL0,Q0,T0,FACT,ALOGF,
1 DPSI0,DTMAX,DETA0,ETALIM,XSI,XSF
COMMON /NPAR/NPHI,IPAR,NP,NR,NX,NXS,NS,NSPEC,NS1,NS2,NTAU0,NETA0,
1 NAMB,NCASE,ICASE,IFAN
COMMON /GEOM/APF,PAI,PAI2,W,SW,CW,BETA,SBETA,CBETA,PSI1,SPSI1,
1 CPSI1,PSIF,SPSIF,CPSIF,TPSIF,AK,SK,CK,A0,RF,XF,YF,ZF,
2 PHISOF,PHIF,SPHIF,CPHIF,DYMIN,RMIN,XS,DIST,X0,Y0,Z0,
3 DY0,DEG,PSIN,ST1,CT1,OMEGX,OMEGY,OMEGZ,XSV(21)
COMMON /EPSIL/EPSETA,EPST,EPSR
COMMON /EXTREM/TEXT(5),ETAEXT(5),ETAKXT(5),PHIEXT(5),
1 PSIEXT(5),EMEXT(5),FEXT(5),WEXT(5),
2 TMAX(5),ETAKMX(5),ETAMAX(5),PSIMAX(5),
3 EMMAX(5),FMAX(5),
4 RFMAX(5),PHIFMX(5),PHIMAX(5),WMAX(5)
COMMON /SPEC/WAV,XC(5),WC(5),WRC(5),XNAME(5),QFC(5),QDC(5),
1 QU2C(5),FLUXC(5),OMEGA(5),FLXU2C(5),URMSC(5)
AAA=(CW/CPSI1)**2-1.D0
IF(AAA.LT.1.D-10) GO TO 1
TPSI1=SPSI1/CPSI1
AP1=A0+XF*TPSI1
BBB=2.D0*(AP1*CW*TPSI1-SW*APF*(CBETA*CPHIF+SBETA*SPHIF))
CCC=AP1**2-APF**2
DDD=BBB**2-4.D0*AAA*CCC
TLIM=(-BBB+DSQRT(DDD))/(2.D0*AAA)
TLIM=TLIM/RF
RETURN
1 CONTINUE
TLIM=1.D 55
RETURN
END
SUBROUTINE SUMT
IMPLICIT REAL*8(A-H,O-Z)
COMMON /GAMA/G,G1,G2,G3,G4,G5,G6,G7,G8,G9,G10,G11,G12,G13,G14,G15,
1 G16,G17,G18,G19,G20
COMMON /PAR/CO,ENO,EM1,D,SIGMA,TLIM,DR0,EL0,Q0,T0,FACT,ALOGF,
1 DPSI0,DTMAX,DETA0,ETALIM,XSI,XSF
COMMON /NPAR/NPHI,IPAR,NP,NR,NX,NXS,NS,NSPEC,NS1,NS2,NTAU0,NETA0,
1 NAMB,NCASE,ICASE,IFAN
COMMON /GEOM/APF,PAI,PAI2,W,SW,CW,BETA,SBETA,CBETA,PSI1,SPSI1,

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1      CPSI1,PSIF,SPSIF,CPSIF,TPSIF,AK,SK,CK,A0,RF,XF,YF,ZF,AMB0577
2      PHISOF,PHIF,SPHIF,CPHIF,DYMIN,RMIN,XS,DIST,X0,Y0,Z0,AMB0578
3      DY0,DEG,PSIN,ST1,CT1,OMEGX,OMEGY,OMEGZ,XSV(21)AMB0579
COMMON /EPSIL/EPSETA,EPST,EPSRAMB0580
COMMON /EXTREM/TEXT(5),ETAEXT(5),ETAKXT(5),PHIEXT(5),AMB0581
1      PSIEXT(5),EMEXT(5),FEXT(5),WEXT(5),AMB0582
2      TMAX(5),ETAKMX(5),ETAMAX(5),PSIMAX(5),AMB0583
3      EMMAX(5),FMAX(5),AMB0584
4      RFMAX(5),PHIFMX(5),PHIMAX(5),WMAX(5)AMB0585
COMMON /SOFPR/CC,DSUMF,DSUMD,T,ETA,DETA,SUM,DSUM,SUMU,DSUMUAMB0586
COMMON /COUNTS/ICONTC,ICONTT,ICNTOT,ICNTMX,IQTOT(5),ISHAD(5)AMB0587
COMMON /SPEC/WAV,XC(5),WC(5),WRC(5),XNAME(5),QFC(5),QDC(5),AMB0588
1      QU2C(5),FLUXC(5),OMEGA(5),FLXU2C(5),URMSC(5)AMB0589
COMMON /SUMS/SUMF(5),SUMD(5),SUMU2(5)AMB0590
C INTEGRATION OF FLUX ARRIVING ALONG A SINGLE RAYAMB0591
DT=DPSI0AMB0592
PSIN=PSIFAMB0593
ETA1=0.AMB0594
ETA3=0.AMB0595
FETA4=0.AMB0596
FETAU4=0.AMB0597
T=0.AMB0598
SUM=0.AMB0599
SUMU=0.AMB0600
ICONTT=0AMB0601
1 ICONTT=ICONTT+1AMB0602
PSIL=PSINAMB0603
DT2=DT/2.D0AMB0604
DT6=DT/6.D0AMB0605
T1=T+DT2AMB0606
T2=T+DTAMB0607
FETA1=FETA4AMB0608
FETAU1=FETAU4AMB0609
CALL PATHK(T1,ETAK1)AMB0610
CALL FETA(T1,ETA1,ETAK1,GT2,FETA2,FETAU2)AMB0611
FETA3=FETA2AMB0612
FETAU3=FETAU2AMB0613
CALL PATHK(T2,ETAK3)AMB0614
CALL FETA(T2,ETA3,ETAK3,GT4,FETA4,FETAU4)AMB0615
DETA=DT*GT4AMB0616
DSUM=DT6*(FETA1+2.D0*(FETA2+FETA3)+FETA4)AMB0617
DSUMU=DT6*(FETAU1+2.D0*(FETAU2+FETAU3)+FETAU4)AMB0618
T=T+DTAMB0619
ETA=ETA3AMB0620
ETAK=ETAK3AMB0621
SUM=SUM+DSUMAMB0622
SUMU=SUMU+DSUMUAMB0623
IF(FEXT(NS).GT.FETA4) GO TO 10AMB0624
FEXT(NS)=FETA4AMB0625
TEXT(NS)=TAMB0626
ETAEXT(NS)=ETAAMB0627
ETAKXT(NS)=ETAKAMB0628
10 CONTINUEAMB0629
C STEP CONTROL (DT)AMB0630
CALL FAN(T,PSI,PHI)AMB0631
IF(PSI.LT.PSIF-1.D-10) CALL SOF('PSI.LT.PSIF')AMB0632
IF(PSI.GT.PSI1) PSI=PSI1AMB0633
PSIN=PSIAMB0634
DPSI=PSIN-PSILAMB0635
DTP=DT*(DPSI0/(DPSI+1.D-10))AMB0636
DTE=DT*(DETA0/(DETA+1.D-10))AMB0637
DT1=1.2D0*DTAMB0638
DT=DMIN1(DTP,DTE,DT1,DTMAX)AMB0639
IF(DT.LE.0.) CALL SOF('COMPUTED DT NEGATIVE')AMB0640
15 CONTINUEAMB0641
IF(IPAR.LT.1)AMB0642
1PRINT 111,NR,NP,T,PSI*DEG,PHI*DEG,ETA,ETAK,SUM,DSUM/(SUM+1.D-20)AMB0643
111 FORMAT(1X,'NR,NP,T,PSI,PHI=',2I3,3D12.3/AMB0644
1 1X,'ETA,ETAK,SUM,ERRR=',4D12.3)AMB0645
IF(ICONTT.GT.ICNTMX)AMB0646
1CALL SOF('ICONTT TOO LARGE')AMB0647
IF(ICONTT.LE.2) GO TO 1AMB0648

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IF(ETA+ETAK.GT.ETALIM) GO TO 100
IF(T.GT.50.D0 .OR. T*RF.GT.A0) GO TO 100
IF(SUM.EQ.0.) GO TO 1
ERR=(DSUM/SUM)/DT
100 IF(ERR.GT.EPST) GO TO 1
CONTINUE
SUM=SUM*ELO
SUMU=SUMU*ELO
RETURN
END
SUBROUTINE FETA(T,ETAIK,ETAK,GT,FET,FETU2)
IMPLICIT REAL*8(A-H,O-Z)
REAL*8 MU1,MU2
COMMON /GAMA/G,G1,G2,G3,G4,G5,G6,G7,G8,G9,G10,G11,G12,G13,G14,G15,
1 G16,G17,G18,G19,G20
COMMON /PAR/C0,EN0,EM1,D,SIGMA,TLIM,DR0,ELO,Q0,T0,FACT,ALOGF,
1 DPSI0,DTMAX,DETA0,ETALIM,XSI,XSF
COMMON /NPAR/NPHI,IPAR,NP,NR,NX,NXS,NS,NSPEC,NS1,NS2,NTAU0,NETA0,
1 NAMB,NCASE,ICASE,IFAN
COMMON /GEOM/APF,PAI,PAI2,W,SW,CW,BETA,SBETA,CBETA,PSI1,SPSI1,
1 CPSI1,PSIF,SPSIF,CPSIF,TPSIF,AK,SK,CK,A0,RF,XF,YF,ZF,
2 PHISOF,PHIF,SPHIF,CPHIF,DYMIN,RMIN,XS,DIST,X0,Y0,Z0,
3 DY0,DEG,PSIN,ST1,CT1,OMEGX,OMEGY,OMEGZ,XSV(21)
COMMON /EPSIL/EPSETA,EPST,EPSR
COMMON /EXTREM/TEXT(5),ETAEXT(5),ETAKXT(5),PHIEXT(5),
1 PSIEXT(5),EMEXT(5),FEXT(5),WEXT(5),
2 TMAX(5),ETAKMX(5),ETAMAX(5),PSIMAX(5),
3 EMMAX(5),FMAX(5),
4 RFMAX(5),PHIFMX(5),PHIMAX(5),WMAX(5)
COMMON /SPEC/WAV,XC(5),WC(5),WRC(5),XNAME(5),QFC(5),QDC(5),
1 QUZC(5),FLUXC(5),OMEGA(5),FLXU2C(5),URMSC(5)
COMMON /AMBIEN/ENA,UA,PSIA,PHIA,HA(3),WA(3),
1 UAX,UAY,UAZ,AA,BA,CA,RA,XA,YA,ZA,SHADOW
LOGICAL SHADOW
COMMON /NAGESH/PIK,UIK,UIKX,UIKY,UIKZ
ETAIK=0.
IF(SHADOW) GO TO 1
K=1
I=NS
CALL FAN(T,PSI,PHI)
IF(PSI.LT.PSIF-1.D-10) CALL SOF('PSI.LT.PSIF')
IF(PSI.GT.PSI1) PSI=PSI1
PSI0=PSI
CALL MATCH(T,PSI0,EM,TETA)
SPSI=DSIN(PSI)
CPSI=DCOS(PSI)
SPHI=DSIN(PHI)
CPHI=DCOS(PHI)
ST=DSIN(TETA)
CT=DCOS(TETA)
GOREM=1.D0+G1*EM**2
TERMN=GOREM**G6
U=EM*C0/DSQRT(GOREM)
UX=U*CT
UY=U*ST*CPHI
UZ=U*ST*SPHI
C COLLISION
MU1=WC(I)/(WC(I)+WA(K))
MU2=1.D0-MU1
UMX=MU1*UX+MU2*UAX
UMY=MU1*UY+MU2*UAY
UMZ=MU1*UZ+MU2*UAZ
DOTUM=OMEGX*UMX+OMEGY*UMY+OMEGZ*UMZ
URX=UX-UAX
URY=UY-UAY
URZ=UZ-UAZ
UR=DSQRT(URX**2+URY**2+URZ**2)
DET=DOTUM**2+(MU2*UR)**2-(UMX**2+UMY**2+UMZ**2)
IF(DET.LT.0.) GO TO 1
DET1=DSQRT(DET)
UIK1=-DOTUM+DET1
UIK2=-DOTUM-DET1

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AMB0649
AMB0650
AMB0651
AMB0652
AMB0653
AMB0654
AMB0655
AMB0656
AMB0657
AMB0658
AMB0659
AMB0660
AMB0661
AMB0662
AMB0663
AMB0664
AMB0665
AMB0666
AMB0667
AMB0668
AMB0669
AMB0670
AMB0671
AMB0672
AMB0673
AMB0674
AMB0675
AMB0676
AMB0677
AMB0678
AMB0679
AMB0680
AMB0681
AMB0682
AMB0683
AMB0684
AMB0685
AMB0686
AMB0687
AMB0688
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AMB0691
AMB0692
AMB0693
AMB0694
AMB0695
AMB0696
AMB0697
AMB0698
AMB0699
AMB0700
AMB0701
AMB0702
AMB0703
AMB0704
AMB0705
AMB0706
AMB0707
AMB0708
AMB0709
AMB0710
AMB0711
AMB0712
AMB0713
AMB0714
AMB0715
AMB0716
AMB0717
AMB0718
AMB0719
AMB0720

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IF(UIK2.GT.0.) CALL SOF('DOUBLE COLLISION OPTION NOT PROGRAMMED
1 YET')
UIK=UIK1
IF(UIK.LE.0.) GO TO 1
UIKX=-OMEGX*UIK
UIKY=-OMEGY*UIK
UIKZ=-OMEGZ*UIK
CDEL=(DOTUM+UIK)/(MU2*UR)
IF(CDEL.LE.0.) CALL SOF('CDEL NEGATIVE NOT PROGRAMMED YET')
IF(CDEL-1.D-10.GT.1.D0)
1CALL SOF('CDEL (COS(Delta)) CANNOT BE GT.1.')
PIK=(UIK/(MU2*UR))*2/(4.D0*PAI*CDEL)
IF (PIK.LT.0.) CALL SOF('PIK.LT.0')
FET=(UR/UA)*PIK/TERMN
UREL=DSQRT((UX-UIKX)**2+(UY-UIKY)**2+(UZ-UIKZ)**2)
GT=EL0*(UREL/UIK)/TERMN
CALL PATHIK(T,ETAIK)
POWER=ETAIK+ETAK-ALOGF
EFACT=0.
IF(POWER.LT.60.D0)EFACT=DEXP(-POWER)
FET=FET*EFACT
FETU2=FET*UIK**2
IF(EM.LT.0.) CALL SOF('EM.LT.0')
FETU2=FET*EM
RETURN
1 CONTINUE
FET=0.
FETU2=0.
GT=0.
RETURN
END
SUBROUTINE PATHIK(TC,ETAIK)
IMPLICIT REAL*8(A-H,O-Z)
REAL*8 MU1,MU2
COMMON /GAMA/G,G1,G2,G3,G4,G5,G6,G7,G8,G9,G10,G11,G12,G13,G14,G15,
1 G16,G17,G18,G19,G20
COMMON /PAR/CO,ENO,EM1,D,SIGMA,TLIM,DR0,EL0,Q0,T0,FACT,ALOGF,
1 DPSI0,DTMAX,DETA0,ETALIM,XSI,XSF
COMMON /NPAR/NPHI,IPAR,NP,NR,NX,NXS,NS,NSPEC,NS1,NS2,NTAU0,NETA0,
1 NAMB,NCASE,ICASE,IFAN
COMMON /GEOM/APF,PAI,PAI2,W,SW,CW,BETA,SBETA,CBETA,PSI1,SPSI1,
1 CPSI1,PSIF,SPSIF,CPSIF,TPSIF,AK,SK,CK,A0,RF,XF,YF,ZF,
2 PHISOF,PHIF,SPHIF,CPHIF,DYMIN,RMIN,XS,DIST,X0,Y0,Z0,
3 DY0,DEG,PSIN,ST1,CT1,OMEGX,OMEGY,OMEGZ,XSV(21)
COMMON /EPSIL/EPSETA,EPST,EPSR
COMMON /EXTREM/TEXT(5),ETAEXT(5),ETAKXT(5),PHIEXT(5),
1 PSIEXT(5),EMEXT(5),FEXT(5),WEXT(5),
2 TMAX(5),ETAKMX(5),ETAMAX(5),PSIMAX(5),
3 EMMAX(5),FMAX(5),
4 RFMAX(5),PHIFMX(5),PHIMAX(5),WMAX(5)
COMMON /SPEC/WAV,XC(5),WC(5),WRC(5),XNAME(5),QFC(5),QDC(5),
1 QU2C(5),FLUXC(5),OMEGA(5),FLXU2C(5),URMSC(5)
COMMON /AMBIEN/ENA,UA,PSIA,PHIA,HA(3),WA(3),
1 UAX,UAY,UAZ,AA,BA,CA,RA,XA,YA,ZA,SHADOW
LOGICAL SHADOW
NETA=NETA0
DT=TC/DBLE(NETA)
DT2=DT/2.D0
DT6=DT/6.D0
GT4=0.
T=0.
ETA=0.
IT=0
1 IT=IT+1
T1=T+DT2
T2=T+DT
GT1=GT4
CALL FT(T1,GT2)
GT3=GT2
CALL FT(T2,GT4)
DETA=DT6*(GT1+2.D0*(GT2+GT3)+GT4)
T=T+DT
AMB0721
AMB0722
AMB0723
AMB0724
AMB0725
AMB0726
AMB0727
AMB0728
AMB0729
AMB0730
AMB0731
AMB0732
AMB0733
AMB0734
AMB0735
AMB0736
AMB0737
AMB0738
AMB0739
AMB0740
AMB0741
AMB0742
AMB0743
AMB0744
AMB0745
AMB0746
AMB0747
AMB0748
AMB0749
AMB0750
AMB0751
AMB0752
AMB0753
AMB0754
AMB0755
AMB0756
AMB0757
AMB0758
AMB0759
AMB0760
AMB0761
AMB0762
AMB0763
AMB0764
AMB0765
AMB0766
AMB0767
AMB0768
AMB0769
AMB0770
AMB0771
AMB0772
AMB0773
AMB0774
AMB0775
AMB0776
AMB0777
AMB0778
AMB0779
AMB0780
AMB0781
AMB0782
AMB0783
AMB0784
AMB0785
AMB0786
AMB0787
AMB0788
AMB0789
AMB0790
AMB0791
AMB0792

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ETA=ETA+DETA
IF(IT.LT.NETA) GO TO 1
ETAIK=ETA
RETURN
END
SUBROUTINE FT(T,GT)
IMPLICIT REAL*8(A-H,O-Z)
REAL*8 MU1,MU2
COMMON /GAMA/G,G1,G2,G3,G4,G5,G6,G7,G8,G9,G10,G11,G12,G13,G14,G15,
1 G16,G17,G18,G19,G20
COMMON /PAR/CO,ENO,EM1,D,SIGMA,TLIM,DR0,ELO,Q0,T0,FACT,ALOGF,
1 DPSIO,DTMAX,DETA0,ETALIM,XSI,XSF
COMMON /NPAR/NPHI,IPAR,NP,NR,NX,NXS,NS,NSPEC,NS1,NS2,NTAU0,NETA0,
1 NAMB,NCASE,ICASE,IFAN
COMMON /GEOM/APF,PAI,PAI2,W,SW,CW,BETA,SBETA,CBETA,PSI1,SPSI1,
1 CPSI1,PSIF,SPSIF,CPSIF,TPSIF,AK,SK,CK,A0,RF,XF,YF,ZF,
2 PHISOF,PHIF,SPHIF,CPHIF,DYMIN,RMIN,XS,DIST,X0,Y0,Z0,
3 DY0,DEG,PSIN,ST1,CT1,OMEGX,OMEGY,OMEGZ,XSV(21)
COMMON /EPSIL/EPSETA,EPST,EPSR
COMMON /EXTREM/TEXT(5),ETAEXT(5),ETAKXT(5),PHIEXT(5),
1 PSIEXT(5),EMEXT(5),FEXT(5),WEXT(5),
2 TMAX(5),ETAKMX(5),ETAMAX(5),PSIMAX(5),
3 EMMAX(5),FMAX(5),
4 RFMAX(5),PHIFMX(5),PHIMAX(5),WMAX(5)
COMMON /SPEC/WAV,XC(5),WC(5),WRC(5),XNAME(5),QFC(5),QDC(5),
1 QU2C(5),FLUXC(5),OMEGA(5),FLXU2C(5),URMSC(5)
COMMON /AMBIEN/ENA,UA,PSIA,PHIA,HA(3),WA(3),
1 UAX,UAY,UAZ,AA,BA,CA,RA,XA,YA,ZA,SHADOW
LOGICAL SHADOW
COMMON /NAGESH/PIK,UIK,UIKX,UIKY,UIKZ
K=1
I=NS
CALL FAN(T,PSI,PHI)
IF(PSI.LT.PSIF-1.D-10) CALL SOF('PSI.LT.PSIF')
IF(PSI.GT.PSI1) PSI=PSI1
PSIO=PSI
CALL MATCH(T,PSIO,EM,TETA)
SPSI=DSIN(PSI)
CPSI=DCOS(PSI)
SPHI=DSIN(PHI)
CPHI=DCOS(PHI)
ST=DSIN(TETA)
CT=DCOS(TETA)
GOREM=1.D0+G1*EM**2
TERMN=GOREM**G6
U=EM*CO/DSQRT(GOREM)
UX=U*CT
UY=U*ST*CPHI
UZ=U*ST*SPHI
UREL=DSQRT((UX-UIKX)**2+(UY-UIKY)**2+(UZ-UIKZ)**2)
GT=ELO*(UREL/UIK)/TERMN
RETURN
END
SUBROUTINE PATHK(T,ETAK)
IMPLICIT REAL*8(A-H,O-Z)
COMMON /GAMA/G,G1,G2,G3,G4,G5,G6,G7,G8,G9,G10,G11,G12,G13,G14,G15,
1 G16,G17,G18,G19,G20
COMMON /PAR/CO,ENO,EM1,D,SIGMA,TLIM,DR0,ELO,Q0,T0,FACT,ALOGF,
1 DPSIO,DTMAX,DETA0,ETALIM,XSI,XSF
COMMON /NPAR/NPHI,IPAR,NP,NR,NX,NXS,NS,NSPEC,NS1,NS2,NTAU0,NETA0,
1 NAMB,NCASE,ICASE,IFAN
COMMON /GEOM/APF,PAI,PAI2,W,SW,CW,BETA,SBETA,CBETA,PSI1,SPSI1,
1 CPSI1,PSIF,SPSIF,CPSIF,TPSIF,AK,SK,CK,A0,RF,XF,YF,ZF,
2 PHISOF,PHIF,SPHIF,CPHIF,DYMIN,RMIN,XS,DIST,X0,Y0,Z0,
3 DY0,DEG,PSIN,ST1,CT1,OMEGX,OMEGY,OMEGZ,XSV(21)
COMMON /EPSIL/EPSETA,EPST,EPSR
COMMON /EXTREM/TEXT(5),ETAEXT(5),ETAKXT(5),PHIEXT(5),
1 PSIEXT(5),EMEXT(5),FEXT(5),WEXT(5),
2 TMAX(5),ETAKMX(5),ETAMAX(5),PSIMAX(5),
3 EMMAX(5),FMAX(5),
4 RFMAX(5),PHIFMX(5),PHIMAX(5),WMAX(5)
COMMON /COUNTS/ICONTC,ICONTT,ICNTOT,ICNTMX,IQTOT(5),ISHAD(5)

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AMB0793
 AMB0794
 AMB0795
 AMB0796
 AMB0797
 AMB0798
 AMB0799
 AMB0800
 AMB0801
 AMB0802
 AMB0803
 AMB0804
 AMB0805
 AMB0806
 AMB0807
 AMB0808
 AMB0809
 AMB0810
 AMB0811
 AMB0812
 AMB0813
 AMB0814
 AMB0815
 AMB0816
 AMB0817
 AMB0818
 AMB0819
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 AMB0861
 AMB0862
 AMB0863
 AMB0864

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COMMON /AMBIEN/ENA,UA,PSIA,PHIA,HA(3),WA(3),
1      UAX,UAY,UAZ,AA,BA,CA,RA,XA,YA,ZA,SHADOW
LOGICAL SHADOW
ETAK=0.
C DETERMINE POINT OF ENTRY OF AMBIENT TRAJECTORY TO FAN
TRF=T*RF
XC=XF+TRF*OMEGX
YC=YF+TRF*OMEGY
ZC=ZF+TRF*OMEGZ
C CHECK SHADOW
SHADOW=.FALSE.
EVER=BA**2+CA**2
DETS=EVER*A0**2-(BA*ZC-CA*YC)**2
IF(DETS.LE.0.) GO TO 2
DETS1=DSQRT(DETS)
TAU1=(-(BA*YC+CA*ZC)+DETS1)/EVER
IF(TAU1.GT.0.) SHADOW=.TRUE.
2 CONTINUE
IF(SHADOW) GO TO 10
EVER1=A0+XC*TPSIF
EVER2=BA**2+CA**2-(AA*TPSIF)**2
EVER3=BA*YC+CA*ZC-AA*EVER1*TPSIF
DET=EVER3**2-EVER2*(YC**2+ZC**2-EVER1**2)
IF(DET.LE.0.)
1CALL SOF('NO INTERSECTION OF AMB. TRAJ. WITH LIMITING CONE')
DET1=DSQRT(DET)
TAUP=(-EVER3+DET1)/EVER2
TAUM=(-EVER3-DET1)/EVER2
IF(TAUP.GT.0. .AND. TAUM.GT.0.)
1CALL SOF('TWO POSITIVE INTERSECTIONS WITH LIMITING CONE.NOT PERMIT
1 IN THIS VERSION')
TAUF=DMAX1(TAUP,TAUM)
IF(TAUF.LE.0.)
1CALL SOF('NO POSITIVE INTERSECTION WITH LIMITING CONE')
XA=XC+TAUF*AA
YA=YC+TAUF*BA
ZA=ZC+TAUF*CA
RA=DSQRT(XA**2+(DSQRT(YA**2+ZA**2)-A0)**2)
TAUF=TAUF/RA
NTAU=NTAU0
DTAU=TAUF/DBLE(NTAU)
ETAK=0.
TAU=0.
DTAU2=DTAU/2.DO
DTAU6=DTAU/6.DO
GTAU4=0.
ITAU=0
1 ITAU=ITAU+1
TAU1=TAU+DTAU2
TAU2=TAU+DTAU
GTAU1=GTAU4
CALL FTAU(TAU1,GTAU2)
GTAU3=GTAU2
CALL FTAU(TAU2,GTAU4)
DETAU=DTAU6*(GTAU1+2.DO*(GTAU2+GTAU3)+GTAU4)
TAU=TAU+DTAU
ETAK=ETAK+DETAU
IF(ITAU.LT.NTAU) GO TO 1
ETAK=ETAK*(SIGMA*ENO*RA)
RETURN
10 CONTINUE
ISHAD(NS)=ISHAD(NS)+1
RETURN
END
SUBROUTINE FTAU(TAU,GTAU)
IMPLICIT REAL*8(A-H,O-Z)
COMMON /GAMA/G,G1,G2,G3,G4,G5,G6,G7,G8,G9,G10,G11,G12,G13,G14,G15,
1      G16,G17,G18,G19,G20
COMMON /PAR/CO,ENO,EM1,D,SIGMA,TLIM,DR0,EL0,Q0,T0,FACT,ALOGF,
1      DPSIO,DTMAX,DETA0,ETALIM,XSI,XSF
COMMON /NPAR/NPHI,IPAR,NP,NR,NX,NXS,NS,NSPEC,NS1,NS2,NTAU0,NETA0,
1      NAMB,NCASE,ICASE,IFAN

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AMB0865
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AMB0877
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AMB0936

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COMMON /GEOM/APF,PAI,PAI2,W,SW,CW,BETA,SBETA,CBETA,PSI1,SPSI1,      AMB0937
1   CPSI1,PSIF,SPSIF,CPSIF,TPSIF,AK,SK,CK,A0,RF,XF,YF,ZF,          AMB0938
2   PHISOF,PHIF,SPHIF,CPHIF,DYMIN,RMIN,XS,DIST,X0,Y0,Z0,          AMB0939
3   DY0,DEG,PSIN,ST1,CT1,OMEGX,OMEGY,OMEGZ,XSV(21)                AMB0940
COMMON /EPSIL/EPSETA,EPST,EPSR                                      AMB0941
COMMON /EXTREM/TEXT(5),ETAEXT(5),ETAKXT(5),PHIEXT(5),              AMB0942
1   PSIEXT(5),EMEXT(5),FEXT(5),WEXT(5),                             AMB0943
2   TMAX(5),ETAKMX(5),ETAMAX(5),PSIMAX(5),                         AMB0944
3   EMMAX(5),FMAX(5),                                               AMB0945
4   RFMAX(5),PHIFMX(5),PHIMAX(5),WMAX(5)                           AMB0946
COMMON /SPEC/WAV,XC(5),WC(5),WRC(5),XNAME(5),QFC(5),QDC(5),       AMB0947
1   QU2C(5),FLUXC(5),OMEGA(5),FLXU2C(5),URMSC(5)                  AMB0948
COMMON /AMBIEN/ENA,UA,PSIA,PHIA,HA(3),WA(3),                       AMB0949
1   UAX,UAY,UAZ,AA,BA,CA,RA,XA,YA,ZA,SHADOW                        AMB0950
LOGICAL SHADOW                                                      AMB0951
CALL FANT(TAU,PSI,PHI)                                             AMB0952
IF(PSI.LT.PSIF-1.D-10) CALL SOF('PSI.LT.PSIF')                    AMB0953
IF(PSI.GT.PSI1) PSI=PSI1                                           AMB0954
PSI0=PSI                                                            AMB0955
CALL MATCH(T,PSI0,EM,TETA)                                         AMB0956
SPSI=DSIN(PSI)                                                      AMB0957
CPSI=DCOS(PSI)                                                      AMB0958
SPHI=DSIN(PHI)                                                      AMB0959
CPHI=DCOS(PHI)                                                      AMB0960
ST=DSIN(TETA)                                                       AMB0961
CT=DCOS(TETA)                                                       AMB0962
GOREM=1.D+G1*EM**2                                                 AMB0963
TERMN=GOREM**G6                                                    AMB0964
U=EM*CO/DSQRT(GOREM)                                               AMB0965
UREL=DSQRT((CT*U-UAX)**2+(ST*CPHI*U-UAY)**2+(ST*SPHI*U-UAZ)**2) AMB0966
GTAU=UREL/(UA*TERMN)                                              AMB0967
RETURN                                                                AMB0968
END                                                                    AMB0969
SUBROUTINE FAN(T,PSI,PHI)                                           AMB0970
IMPLICIT REAL*8(A-H,O-Z)                                           AMB0971
COMMON /PAR/CO,ENO,EM1,D,SIGMA,TLIM,DR0,ELO,Q0,T0,FACT,ALOGF,     AMB0972
1   DPSI0,DTMAX,DETA0,ETALIM,XSI,XSF                               AMB0973
COMMON /GEOM/APF,PAI,PAI2,W,SW,CW,BETA,SBETA,CBETA,PSI1,SPSI1,   AMB0974
1   CPSI1,PSIF,SPSIF,CPSIF,TPSIF,AK,SK,CK,A0,RF,XF,YF,ZF,        AMB0975
2   PHISOF,PHIF,SPHIF,CPHIF,DYMIN,RMIN,XS,DIST,X0,Y0,Z0,        AMB0976
3   DY0,DEG,PSIN,ST1,CT1,OMEGX,OMEGY,OMEGZ,XSV(21)              AMB0977
COMMON /POINT/XP,YP,XCOR,ycor                                       AMB0978
C RING FAN GEOMETRY. FAN CORNER IS AT (0,A0*COS(PHI),A0*SIN(PHI)). AMB0979
C RF -- RADIAL DISTANCE ON LIMITING CHARACTERISTIC OF POINT OF    AMB0980
C ENTRANCE OF RAY.                                               AMB0981
C DIRECTION COSINES OF RAY: OMEGX,OMEGY,OMEGZ                      AMB0982
TRF=T*RF                                                            AMB0983
X=XF+TRF*OMEGX                                                      AMB0984
Y=YF+TRF*OMEGY                                                      AMB0985
Z=ZF+TRF*OMEGZ                                                      AMB0986
DY=DSQRT(Y*Y+Z*Z)-A0                                               AMB0987
IF(DABS(DY).LE.1.D-10*A0) DY=1.D-10*A0                             AMB0988
IF(DY.LT.0.)                                                        AMB0989
1CALL SOF('POINT X,Y,X CANNOT BE CLOSER TO X-AXIS THAN RADIUS A0') AMB0990
YY=X/DY                                                             AMB0991
PSI=PAI2-DATAN(YY)                                                  AMB0992
PHI=DATAN(Z/Y)                                                       AMB0993
XP=XCOR+X                                                            AMB0994
YP=A0+DY                                                             AMB0995
RETURN                                                                AMB0996
END                                                                    AMB0997
SUBROUTINE FANT(TAU,PSI,PHI)                                        AMB0998
IMPLICIT REAL*8(A-H,O-Z)                                           AMB0999
COMMON /PAR/CO,ENO,EM1,D,SIGMA,TLIM,DR0,ELO,Q0,T0,FACT,ALOGF,     AMB1000
1   DPSI0,DTMAX,DETA0,ETALIM,XSI,XSF                               AMB1001
COMMON /GEOM/APF,PAI,PAI2,W,SW,CW,BETA,SBETA,CBETA,PSI1,SPSI1,   AMB1002
1   CPSI1,PSIF,SPSIF,CPSIF,TPSIF,AK,SK,CK,A0,RF,XF,YF,ZF,        AMB1003
2   PHISOF,PHIF,SPHIF,CPHIF,DYMIN,RMIN,XS,DIST,X0,Y0,Z0,        AMB1004
3   DY0,DEG,PSIN,ST1,CT1,OMEGX,OMEGY,OMEGZ,XSV(21)              AMB1005
COMMON /AMBIEN/ENA,UA,PSIA,PHIA,HA(3),WA(3),                       AMB1006
1   UAX,UAY,UAZ,AA,BA,CA,RA,XA,YA,ZA,SHADOW                        AMB1007
COMMON /POINT/XP,YP,XCOR,ycor                                       AMB1008

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LOGICAL SHADOW
C RING FAN GEOMETRY. FAN CORNER IS AT (0,A0*COS(PHI),A0*SIN(PHI)).
C RA -- RADIAL DISTANCE ON LIMITING CHARACTERISTIC OF POINT OF
C ENTRANCE OF RAY.
C DIRECTION COSINES OF RAY: -AA,-BA,-CA
TRA=TAU*RA
X=XA-TRA*AA
Y=YA-TRA*BA
Z=ZA-TRA*CA
DY=DSQRT(Y*Y+Z*Z)-A0
IF(DABS(DY).LE.1.D-10*A0) DY=1.D-10*A0
IF(DY.LT.0.)
1CALL SOF('POINT X,Y,X CANNOT BE CLOSER TO X-AXIS THAN RADIUS A0')
YY=X/DY
PSI=PAI2-DATAN(YY)
PHI=DATAN(Z/Y)
XP=XCOR+X
YP=A0+DY
RETURN
END
SUBROUTINE HMSET
C SUBROUTINE NUMBER 20
IMPLICIT REAL*8(A-H,O-Z,$)
REAL*8 KAPAOB,MHINV,MINVO,M,MF,M1,M2,M3,NORM,MEXIT,LAMDOB
COMMON /GAMA/G,G1,G2,G3,G4,G5,G6,G7,G8,G9,G10,G11,G12,G13,G14,G15,
1 G16,G17,G18,G19,G20
COMMON /PAR/C0,EN0,EM1,D,SIGMA,TLIM,DR0,EL0,Q0,TO,FACT,ALOGF,
1 DPSIO,DTMAX,DETA0,ETALIM,XSI,XSF
COMMON /GEOM/APF,PAI,PAI2,W,SW,CW,BETA,SBETA,CBETA,PSI1,SPSI1,
1 CPSI1,PSIF,SPSIF,CPSIF,TPSIF,AK,SK,CK,A0,RF,XF,YF,ZF,
2 PHISOF,PHIF,SPHIF,CPHIF,DYMIN,RMIN,XS,DIST,X0,Y0,Z0,
3 DY0,DEG,PSIN,ST1,CT1,OMEGX,OMEGY,OMEGZ,XSV(21)
COMMON /GRP/DMINV,MHINV(101),HMV(101)
COMMON /IGRP/KHM
C A ROUTINE FOR THE C+ DERIVATIVE DUE TO RING SYMMETRY (GRP).
MEXIT=EM1
KHM=51
IF(KHM.GT.101) CALL SOF('2001')
MINVO=1.D0/MEXIT
DMINV=MINVO/DBLE(KHM-1)
M=MEXIT
SUM=0.
KHM1=KHM-1
DO 1 I=1,KHM1
MF=M
MHINV(I)=MINVO-DBLE(I-1)*DMINV
M=1.D0/MHINV(I)
DM=M-MF
M1=M-DM
M2=M-DM/2.D0
M3=M
CALL MFUNC(M1,F1,ETALF1,TETA1)
CALL MFUNC(M2,F2,ETALF2,TETA2)
CALL MFUNC(M3,F3,ETALF3,TETA3)
SUM=SUM+DM*(F1+4.D0*F2+F3)/6.D0
ETALF=ETALF3
TETA=TETA3
PSI=TETA+DASIN(1.D0/M)
NORM=((3.D0-G)/4.D0)*(M**2-1.D0)**0.75D0/
1 (DSIN(PSI)*(1.D0+G1*M**2)**G14)
HM=SUM*NORM
HMV(I)=HM
GOREM=1.D0+G1*M**2
GOR=M**2-1.D0
DELTOB=0.5D0*DSQRT(GOR)*(1.D0/(MEXIT*ETALF)
1 +DSIN(TETA)/M)/DSIN(PSI)+G15*HM/2.D0
EPSIOB=DELTOB/DSQRT(GOR)-DSIN(TETA)/(M*DSIN(PSI))
KAPAOB=1.D0
IF(DABS(PAI2-TETA).GT.1.D-6)
1KAPAOB=DTAN(TETA)*EPSIOB
LAMDOB=EPSIOB-DELTOB*GOREM/(GOR*DSQRT(GOR))
PRINT 11,I,M,HM,TETA*DEG,PSI*DEG

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11  FORMAT(/1X, '                I,M,HM,TETA,PSI=',I5,5D12.4)      AMB1081
    PRINT 12,DELTOB,EPSIOB*DEG,KAPAOB*DEG,LAMDOB*DEG      AMB1082
12  FORMAT( 1X,'DELTOB,EPSIOB,KAPAOB,LAMDOB=',5X,5D12.4)    AMB1083
1   CONTINUE                                             AMB1084
    MHINV(KHM)=0.                                         AMB1085
    HMOV(KHM)=1.D0                                       AMB1086
    RETURN                                               AMB1087
    END                                                 AMB1088
    SUBROUTINE MFUNC(M,F,ETALF,TETA)                    AMB1089
C   SUBROUTINE NUMBER 21                                AMB1090
    IMPLICIT REAL*8(A-H,O-Z,$)                          AMB1091
    REAL*8 NU,NUFUNC,M,MEXIT,MD,MDD                    AMB1092
    COMMON /GAMA/G,G1,G2,G3,G4,G5,G6,G7,G8,G9,G10,G11,G12,G13,G14,G15, AMB1093
    1   G16,G17,G18,G19,G20                             AMB1094
    COMMON /PAR/CO,ENO,EM1,D,SIGMA,TLIM,DR0,EL0,Q0,TO,FACT,ALOGF, AMB1095
    1   DPSIO,DTMAX,DETA0,ETALIM,XSI,XSF               AMB1096
    COMMON /GEOM/APF,PAI,PAI2,W,SW,CW,BETA,SBETA,CBETA,PSI1,SPSI1, AMB1097
    1   CPSI1,PSIF,SPSIF,CPSIF,TPSIF,AK,SK,CK,A0,RF,XF,YF,ZF, AMB1098
    2   PHISOF,PHIF,SPHIF,CPHIF,DYMIN,RMIN,XS,DIST,XO,YO,ZO, AMB1099
    3   DY0,DEG,PSIN,ST1,CT1,OMEGX,OMEGY,OMEGZ,XSV(21) AMB1100
C   QF(MDD)=1.D0/DSQRT(MDD**2-1.D0)                    AMB1101
    NUFUNC(MD)=-G5*DATAN(G5*QF(MD))+DATAN(QF(MD))      AMB1102
C   MEXIT=EM1                                           AMB1103
    NU=NUFUNC(M)                                       AMB1104
    TETA=NUFUNC(MEXIT)+PAI2-NU                         AMB1105
    GOREM=1.D0+G1*M**2                                 AMB1106
    GOR=M**2-1.D0                                     AMB1107
    F=(M**2)*(GOREM**G13)*DSIN(TETA)/GOR**1.25D0     AMB1108
    GOREM1=1.D0+G1*MEXIT**2                           AMB1109
    GOR1=MEXIT**2-1.D0                                AMB1110
    ETALF=((GOREM/GOREM1)**G14)*((GOR1/GOR)**0.25D0)  AMB1111
    RETURN                                             AMB1112
    END                                               AMB1113
    SUBROUTINE HINTER(M,H)                              AMB1114
C   SUBROUTINE NUMBER 22                                AMB1115
    IMPLICIT REAL*8(A-H,O-Z,$)                          AMB1116
    REAL*8 MINV,M,MEXIT,MHINV                          AMB1117
    COMMON /GAMA/G,G1,G2,G3,G4,G5,G6,G7,G8,G9,G10,G11,G12,G13,G14,G15, AMB1118
    1   G16,G17,G18,G19,G20                             AMB1119
    COMMON /PAR/CO,ENO,EM1,D,SIGMA,TLIM,DR0,EL0,Q0,TO,FACT,ALOGF, AMB1120
    1   DPSIO,DTMAX,DETA0,ETALIM,XSI,XSF               AMB1121
    COMMON /GRP/DMINV,MHINV(101),HMOV(101)              AMB1122
    COMMON /IGRP/KHM                                    AMB1123
C   COMPUTE H(M) BY INTERPOLATION                       AMB1124
    MEXIT=EM1                                           AMB1125
    MINV=1.D0/M                                         AMB1126
    I=KHM-IDINT(MINV/DMINV-1.D-9)-1                   AMB1127
    IF(I.GE.1.AND.I.LT.KHM) GO TO 1                    AMB1128
    PRINT 11,I,KHM,M,MEXIT                              AMB1129
11  FORMAT(/1X,'I,KHM,M,MEXIT=',2I5,2D14.6/)        AMB1130
    CALL SOF('2201')                                    AMB1131
1   CONTINUE                                             AMB1132
    F1=(MINV-MHINV(I+1))/DMINV                         AMB1133
    F2=1.D0-F1                                         AMB1134
    IF(F1.LT.-1.D-9) CALL SOF('2210')                  AMB1135
    IF(F2.LT.-1.D-9) CALL SOF('2211')                  AMB1136
    H=F1*HMOV(I)+F2*HMOV(I+1)                          AMB1137
    RETURN                                             AMB1138
    END                                               AMB1139
    SUBROUTINE MATCH(T,PSIO,MAB,TETAAB)                AMB1140
C   SUBROUTINE NUMBER 23                                AMB1141
    IMPLICIT REAL*8(A-H,O-Z,$)                          AMB1142
    REAL*8 M,MOB,MEXIT,MAB,LAMDOB,KAPAOB              AMB1143
    COMMON /GAMA/G,G1,G2,G3,G4,G5,G6,G7,G8,G9,G10,G11,G12,G13,G14,G15, AMB1144
    1   G16,G17,G18,G19,G20                             AMB1145
    COMMON /PAR/CO,ENO,EM1,D,SIGMA,TLIM,DR0,EL0,Q0,TO,FACT,ALOGF, AMB1146
    1   DPSIO,DTMAX,DETA0,ETALIM,XSI,XSF               AMB1147
    COMMON /NPAR/NPHI,IPAR,NP,NR,NX,NXS,NS,NSPEC,NS1,NS2,NTAU0,NETA0, AMB1148
    1   NAMB,NCASE,ICASE,IFAN                           AMB1149
    COMMON /GEOM/APF,PAI,PAI2,W,SW,CW,BETA,SBETA,CBETA,PSI1,SPSI1, AMB1150
    AMB1151
    AMB1152

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1      CPSI1,PSIF,SPSIF,CPSIF,TPSIF,AK,SK,CK,A0,RF,XF,YF,ZF,AMB1153
2      PHISOF,PHIF,SPHIF,CPHIF,DYMIN,RMIN,XS,DIST,X0,Y0,Z0,AMB1154
3      DY0,DEG,PSIN,ST1,CT1,OMEGX,OMEGY,OMEGZ,XSV(21)AMB1155
COMMON /POINT/XP,YP,XCOR,YCORAMB1156
COMMON /GRP/DMINV,MHINV(101),HMV(101)AMB1157
COMMON /IGRP/KHMAMB1158
MEXIT=EM1AMB1159
GO TO (101,102),IFANAMB1160
101  CONTINUEAMB1161
C  FAN APPROXIMATED AS PLANARAMB1162
MAB=DSQRT(1.DO+G4/DTAN((PSI0-PSIF)/G5)**2)AMB1163
TETAAB=PSI0-DASIN(1.DO/MAB)AMB1164
GO TO 100AMB1165
102  CONTINUEAMB1166
C  COMPUTE MAB FROM THE INVERSE PROBLEM SOLUTIONAMB1167
COTAV=1.DO/DTAN(PSI0)AMB1168
EVY=YP*DLOG(YP/YCOR)/(YP-YCOR)-1.DOAMB1169
PSIN=PSI0AMB1170
DO 1 ITER=1,10AMB1171
PSI=PSINAMB1172
M=DSQRT(1.DO+G4/DTAN((PSI-PSIF)/G5)**2)AMB1173
M=DMAX1(M,MEXIT)AMB1174
CALL HINTER(M,HM)AMB1175
CALL MFUNC(M,F,ETALF,TETA)AMB1176
GOREM=1.DO+G1*M**2AMB1177
GOR=M**2-1.DOAMB1178
DELTOB=0.5D0*DSQRT(GOR)*(1.DO/(MEXIT*ETALF)AMB1179
+DSIN(TETA)/M)/DSIN(PSI)+G15*HM/2.DOAMB1180
EPSIOB=DELTOB/DSQRT(GOR)-DSIN(TETA)/(M*DSIN(PSI))AMB1181
LAMDOB=EPSIOB-DELTOB*GOREM/(GOR*DSQRT(GOR))AMB1182
COTN=COTAV+LAMDOB*EVY/DSIN(PSI)**2AMB1183
PSIN=PAI2-DATAN(COTN)AMB1184
DPSI=PSIN-PSIAMB1185
IF(DABS(DPSI).LT.1.D-6) GO TO 11AMB1186
1  CONTINUEAMB1187
PRINT 12,I,ITER,PSI,PSIN,DPSI,M,XP,YP,TAMB1188
12  FORMAT(/1X,'I,ITER,PSI,PSIN,DPSI,M,XP,YP,T='//AMB1189
1  1X,2I4,7D11.3/)AMB1190
CALL SOF('2301')AMB1191
11  CONTINUEAMB1192
C  USING MOB=M AS COMPUTED FROM THE INVERSE PROBLEM, FIND MAB.AMB1193
MOB=MAMB1194
CALL MFUNC(M,F,ETALF,TETA)AMB1195
PSI=TETA+DASIN(1.DO/M)AMB1196
CALL HINTER(M,HM)AMB1197
GOREM=1.DO+G1*M**2AMB1198
GOR=M**2-1.DOAMB1199
DELTOB=0.5D0*DSQRT(GOR)*(1.DO/(MEXIT*ETALF)AMB1200
+DSIN(TETA)/M)/DSIN(PSI)+G15*HM/2.DOAMB1201
FOB=(G7*GOREM)**G2/MAMB1202
FAB=FOB*(YP/YCOR)**DELTOBAMB1203
CALL AREAF(FAB,MAB)AMB1204
EPSIOB=DELTOB/DSQRT(GOR)-DSIN(TETA)/(M*DSIN(PSI))AMB1205
KAPAOB=1.DOAMB1206
IF(DABS(PAI2-TETA).GT.1.D-8)AMB1207
1KAPAOB=DTAN(TETA)*EPSIOBAMB1208
COSTAB=DCOS(TETA)*(YP/YCOR)**(-KAPAOB)AMB1209
TETAAB=DACOS(COSTAB)AMB1210
100  CONTINUEAMB1211
RETURNAMB1212
ENDAMB1213
SUBROUTINE AREAF(F,M)AMB1214
C  SUBROUTINE NUMBER 24AMB1215
IMPLICIT REAL*8(A-H,O-Z,$)AMB1216
REAL*8 MEXIT,MIN,M,MHINVAMB1217
COMMON /GAMA/G,G1,G2,G3,G4,G5,G6,G7,G8,G9,G10,G11,G12,G13,G14,G15,AMB1218
1  G16,G17,G18,G19,G20AMB1219
COMMON /PAR/CO,ENO,EM1,D,SIGMA,TLIM,DR0,EL0,Q0,T0,FACT,ALOGF,AMB1220
1  DPSIO,DTMAX,DETA0,ETALIM,XSI,XSFAMB1221
COMMON /GRP/DMINV,MHINV(101),HMV(101)AMB1222
COMMON /IGRP/KHMAMB1223
C  COMPUTE MACH NUMBER M FROM AREA RATIO FUNCTION FAMB1224

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C	F=((2/(G+1))*(1+(G-1)*M**2))**((G+1)/(2*(G-1)))/M	AMB1225
C	INITIAL GUESS IS MIN	AMB1226
	MEXIT=EM1	AMB1227
	E1=(F*MEXIT)**(1.D0/G2)/G7	AMB1228
	E2=(E1-1.D0)/G1	AMB1229
	E3=DMAX1(E2,MEXIT**2)	AMB1230
	MIN=DSQRT(E3)	AMB1231
	EMN=MIN	AMB1232
	DO 1 I=1,100	AMB1233
	EMO=EMN	AMB1234
	GOREM=1.D0+G1*EMO**2	AMB1235
	GOR=EMO**2-1.D0	AMB1236
	FO=(G7*GOREM)**G2/EMO	AMB1237
	DF=FO-F	AMB1238
C	PRINT 123,I,EMO,EMN,FO,F,DF,GOR,GOREM	AMB1239
C123	FORMAT(1X,'I,EMO,EMN,FO,F,DF,GOR,GOREM=',I5,7D12.4)	AMB1240
	DFDM=FO*GOR/(EMO*GOREM)	AMB1241
	DMN=DF/DFDM	AMB1242
	EMN=EMO-DMN	AMB1243
	EPSEM=DABS(DMN/EMN)	AMB1244
	IF(EPSEM.LT.1.D-10) GO TO 11	AMB1245
1	CONTINUE	AMB1246
	CALL SOF('2401')	AMB1247
11	CONTINUE	AMB1248
	M=EMN	AMB1249
	RETURN	AMB1250
	END	AMB1251

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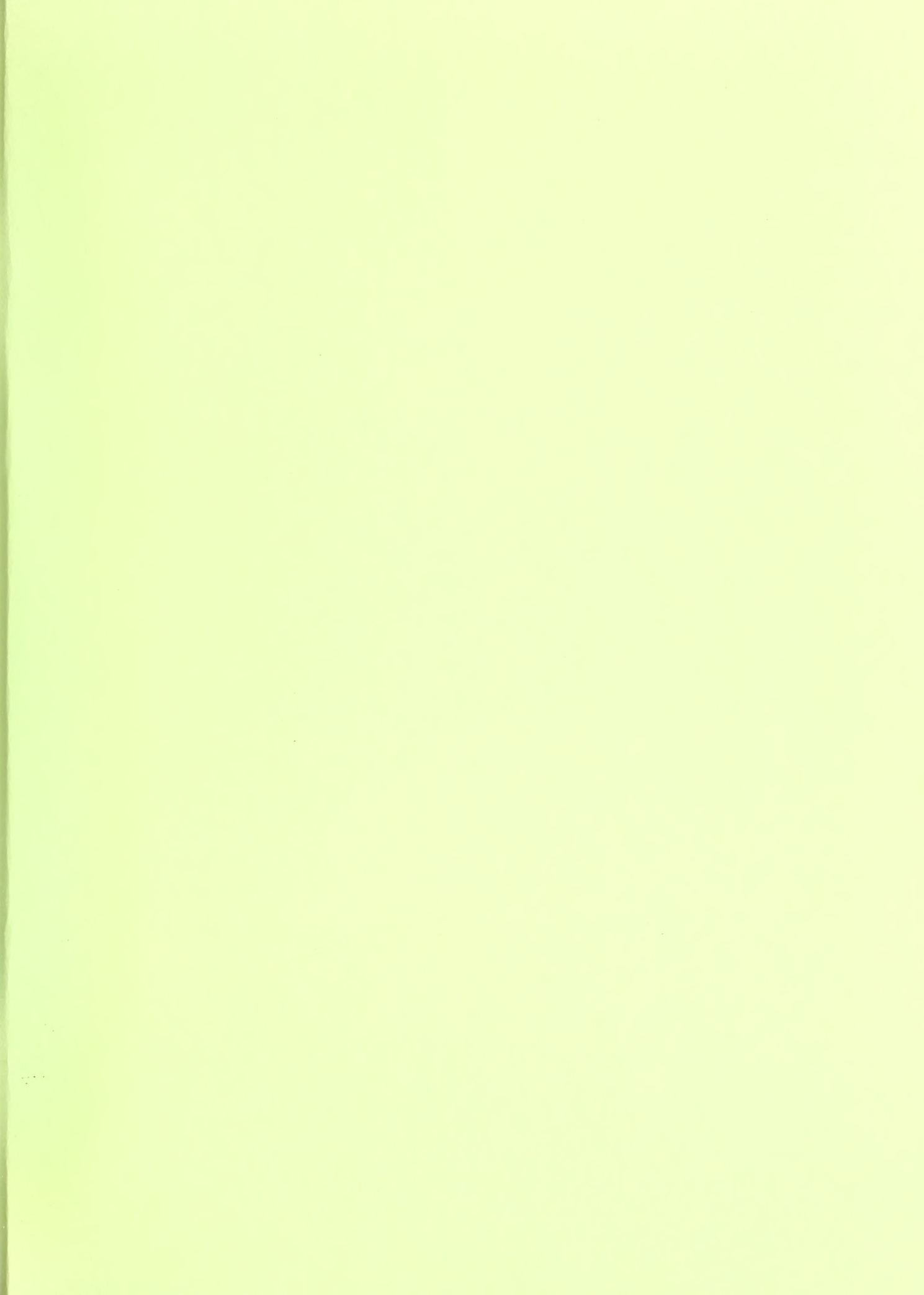
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